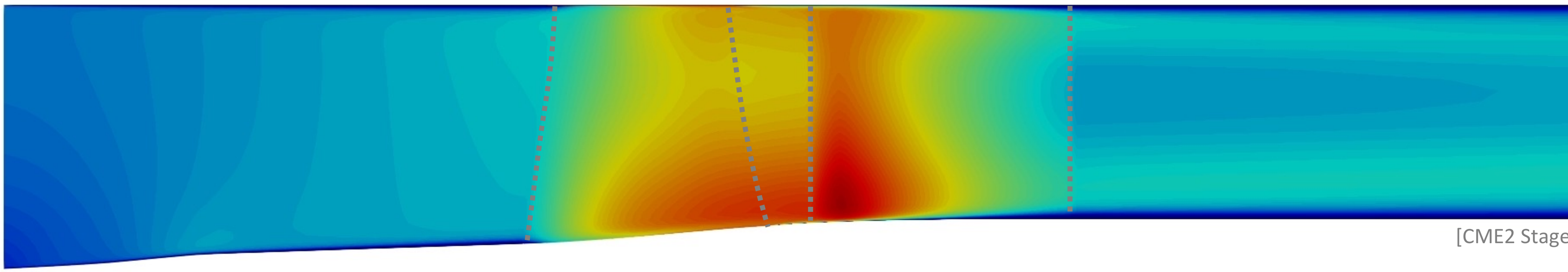
Assessment of geometric variability effects through a viscous through-flow models applied to modern axial-flow compressor blades


[CME2 Stage]

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ETC 2023

LIÈGE université [(1)]

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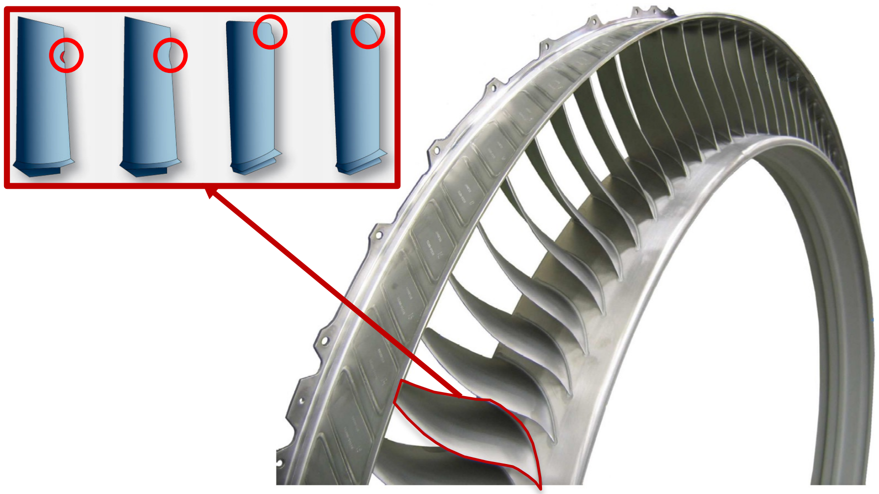
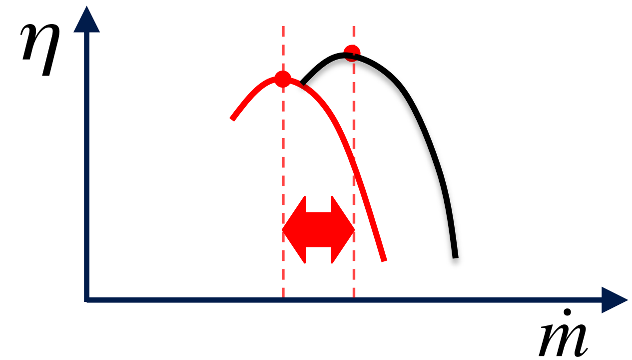
Context

Geometric variability of low-pressure compressor blades



[SAB]

Performance variation



Manufacturing tolerances?

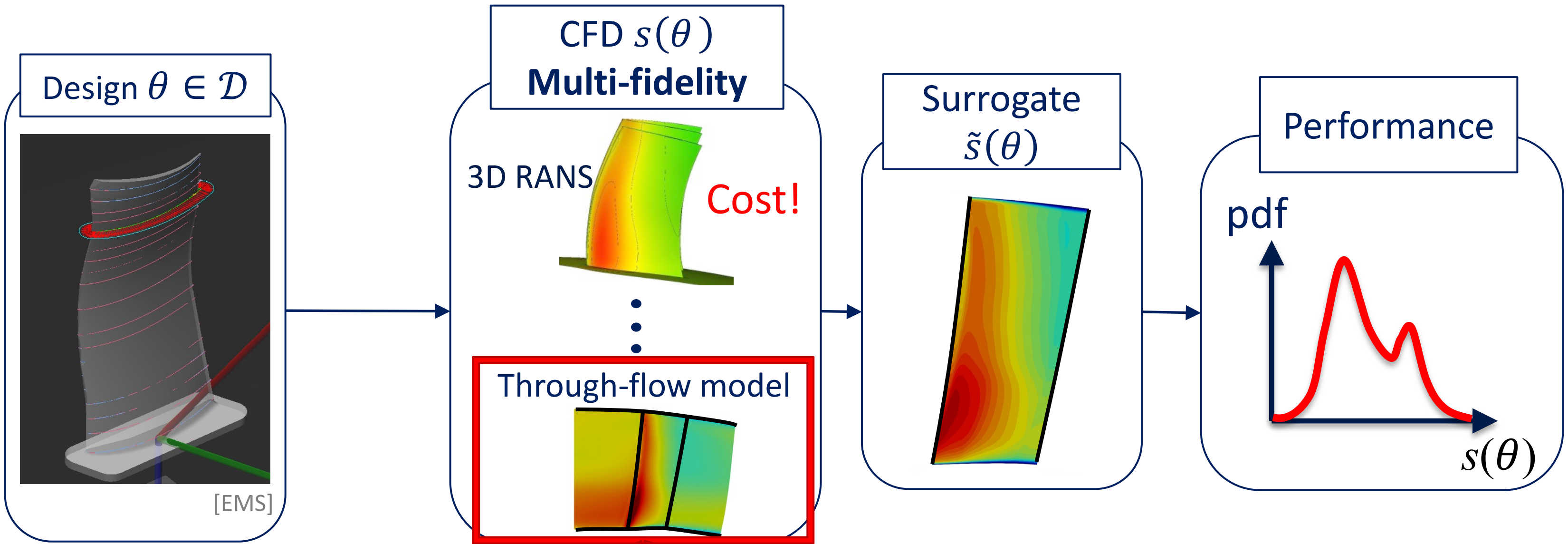
- Need of rigorous/robust definition
- Linked to manufacturing process
- Simplify the treatment of poorly made parts

Trade-off

Cost ⇔ performance

Methodology & objectives

Characterization | **Propagation** | **Qualification**



Able to predict performance?

Through-flow model validation

- Low-fidelity approach
- Choice of model correlations

Able to capture variability effects?

Geometrical variability

- Sensitivity analysis
- (Uncertainty quantification)

Outline

1

Viscous through-flow model

$$\frac{\partial \boldsymbol{U}}{\partial t} + \frac{\partial(\boldsymbol{F}-\boldsymbol{F}_v)}{\partial x} + \frac{\partial(\boldsymbol{G}-\boldsymbol{G}_v)}{\partial r} = \boldsymbol{S}$$

2

Model Assessment

Application to an axial LP compressor

3

Geometrical variability

[SAB]

Outline

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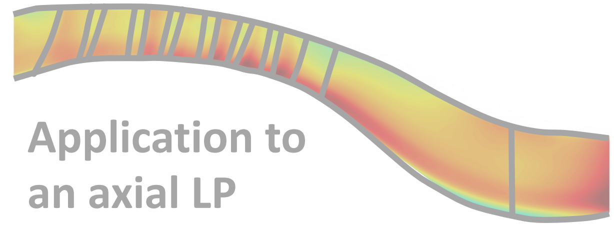
Model Assessment



Application to an axial LP compressor

3

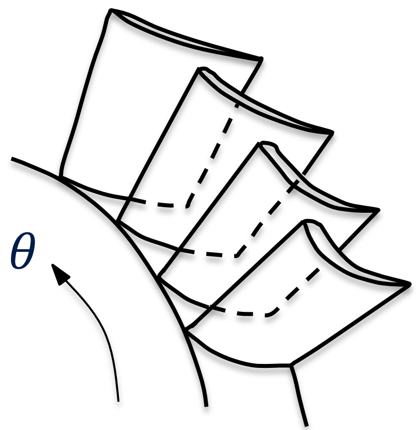
Geometrical variability

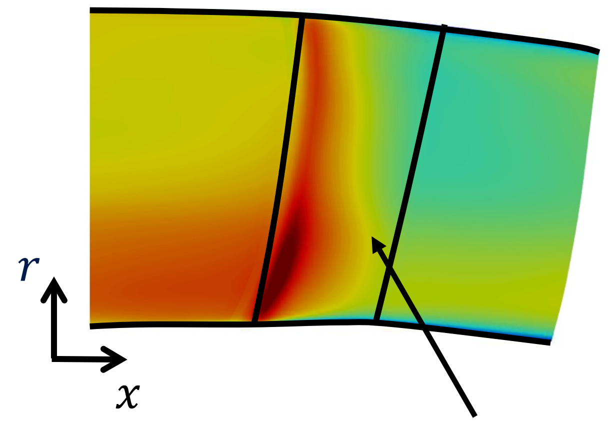
[SAB]

Through-flow model

$$\mathrm{D}_t\boldsymbol{U}(r,\theta,x,t) = \boldsymbol{G}(U,r,\theta,x,t) \xrightarrow{\theta\ -\text{averaging}} \mathrm{D}_t\bar{\boldsymbol{U}}(r,x) = \bar{\boldsymbol{G}}\,(U,r,x)$$

Unclosed!





Implicit presence of the blades

- Azimuthal/pitchwise averaging approach
- Axisymmetric steady flow (meridional plane)
- Empirical correlations
- Low computational cost $\mathcal{O}$(min)

Through-flow formulation

$$\boxed{\mathrm{D}_t\bar{\boldsymbol{U}}(r, x)} = \boxed{\bar{\boldsymbol{G}}(U, r, x)}$$

Governing equations

- Underlying assumptions (NSE, Euler, SLC, …)?
- Consistency?

Closure models

- Exhaustive?
- Assumptions?
- Correlations?

Choices influence **level of empiricism** and determine **error sources**

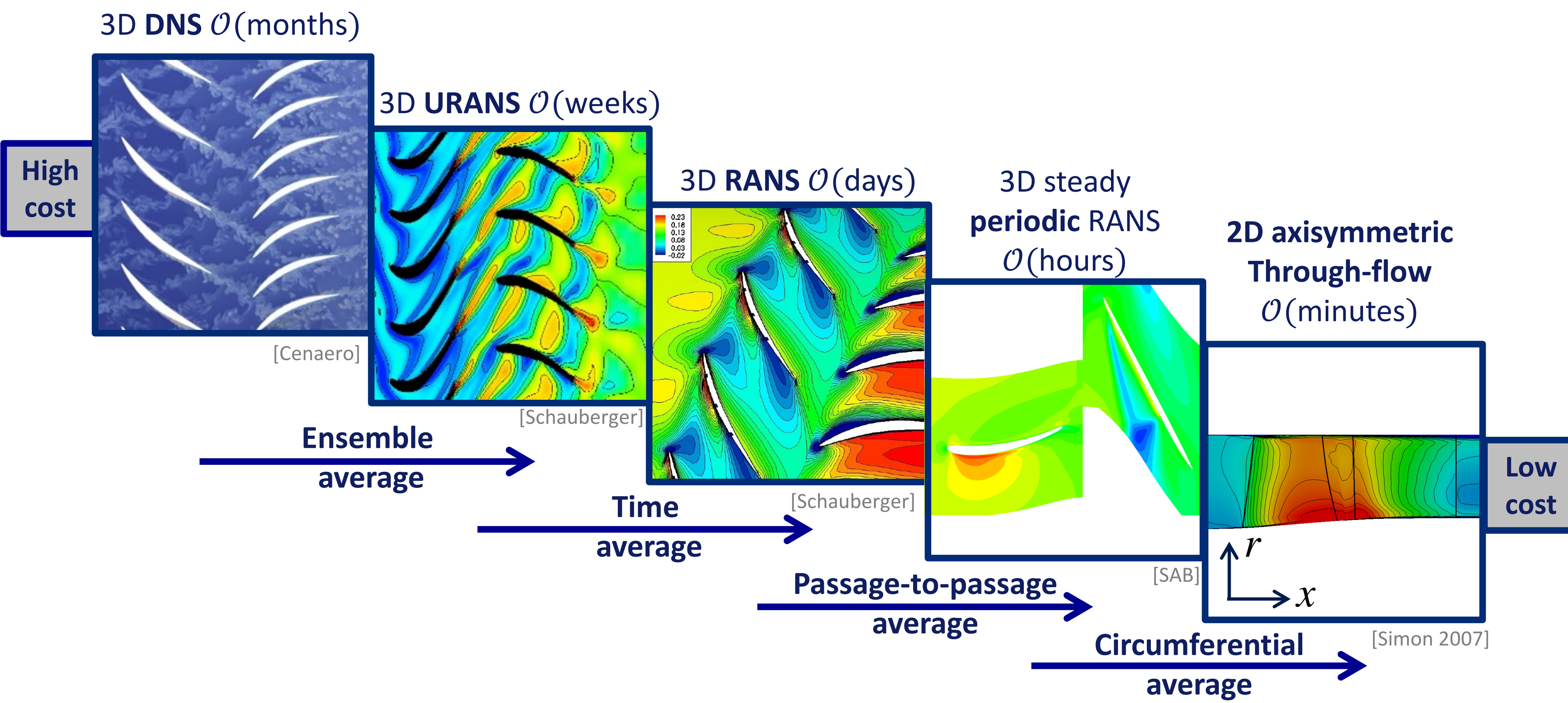
→ **characterize error** in a **rigourous and exhaustive** way

Adamczyck's cascade

$$\mathrm{D}_t\bar{\boldsymbol{U}}(r, x) = \bar{\boldsymbol{G}}(U, r, x)$$

Unclosed!

- **Exact mathematical** formulation of source terms
- **Robust** and **exhaustive** definition of closures
- **NSE-based equations**



Adamczyk's cascade: unclosed terms

$$\mathrm{D}_t\bar{\boldsymbol{U}}(r,x)=\boxed{\bar{\boldsymbol{G}}(U,r,x)}$$

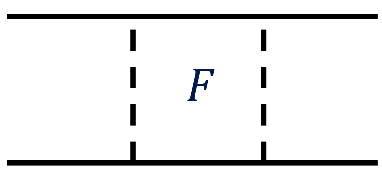
Unclosed terms:

$\overline{\rho V_i' V_i'}$

Non-linear equations

- τ_{reys} ➢ Reynolds
- τ_{uns} ➢ Unsteady
- τ_{ape} ➢ Aperiodic
- τ_{circ} ➢ Circumferential

Stresses



- B_{i} ➢ inviscid
- B_{v} ➢ Viscous

Forces

Empiricism/approximation through model

Relative importance of source terms

$$\mathrm{D}_t\bar{\boldsymbol{U}}(r,x) = \boxed{\bar{\boldsymbol{G}}(U,r,x)}$$

Included
- Reynolds stress
- Inviscid blade force
- Viscous blade force
- Axisymmetric source terms

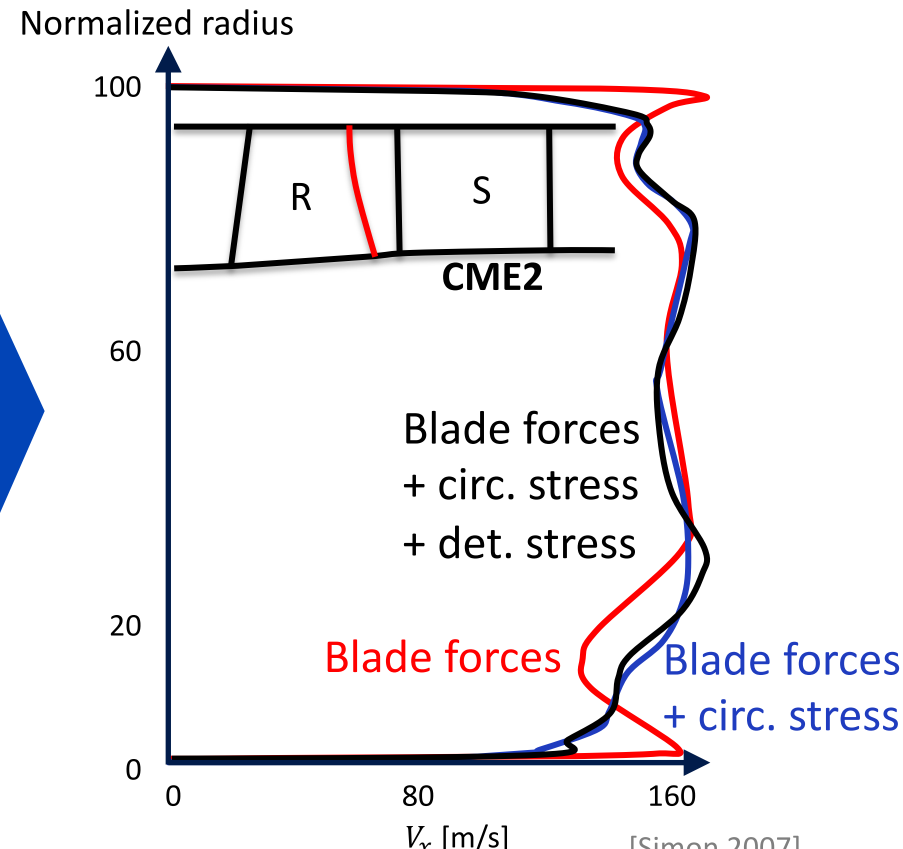
- Reynolds stress
- Inviscid blade force
- Viscous blade force

Major terms

- Circumferential stress
- Unsteady stress

Lower importance

- Aperiodic stress
  → **Generally neglected**



[Simon 2007]

Viscous TF model: closure models

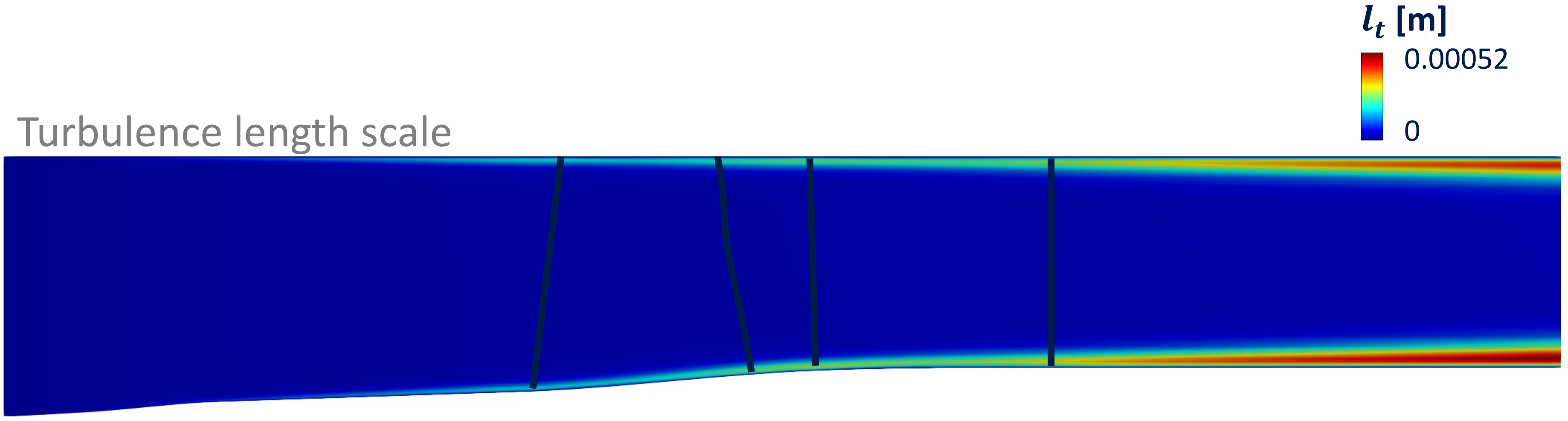
Circumferential averaged Navier-Stokes equations:

$$\mathrm{D}_t \bar{\boldsymbol{U}}(r, x) = \boxed{\bar{\boldsymbol{G}}(U, r, x)}$$

- **Reynolds stress**
- Inviscid blade force
- Viscous blade force
- Axisymmetric source terms

Reynolds stress τ_{reys}: standard **turbulence model** ($k - l$ Smith)

Turbulence length scale



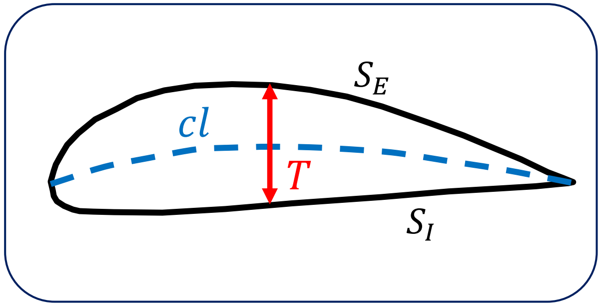
Viscous TF model: closure models

Circumferential averaged Navier-Stokes equations:

$$\mathrm{D}_t\bar{\boldsymbol{U}}(r,x) = \boxed{\bar{\boldsymbol{G}}(U,r,x)}$$

- Reynolds stress
- **Inviscid blade force**
- Viscous blade force
- Axisymmetric source terms

Inviscid blade force decomposition B_{i} **:**



B_{i}

Blade blockage contribution: $B_{\mathrm{i}1} = f\left(\boxed{\frac{P_{S_E}+P_{S_I}}{2}}\right)$

Unknown!

Deflection force: $B_{\mathrm{i}2} = f\left(\boxed{P_{S_E}, P_{S_I}}\right)$

Closure models: blade forces

$$D_t\bar{\boldsymbol{U}}(r, x) = \boxed{\bar{\boldsymbol{G}}(U, r, x)}$$

- Reynolds stress
- **Inviscid blade force**
- **Viscous blade force**
- Axisymmetric source terms

Blade blockage B_{i1}

$$B_{\mathrm{i1}} = B_{\mathrm{i1}}\left(\boxed{\frac{P_{S_E} + P_{S_I}}{2}}\right)$$

Averaged pressure

$$\boldsymbol{S}_{bi1} = \begin{bmatrix} 0 \\ \frac{\boxed{p}}{b}\frac{\partial b}{\partial x} \\ \frac{\boxed{p}}{b}\frac{\partial b}{\partial r} \\ 0 \\ 0 \end{bmatrix}$$

Blockage factor: b

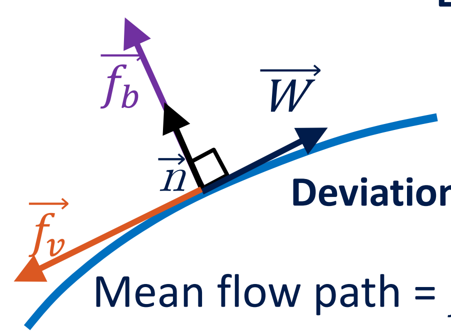
Deflection B_{i2} and viscous force B_{v}

$$B_{\mathrm{i2}} = B_{\mathrm{i2}}\left(\boxed{P_{S_E}, P_{S_I}}\right) \qquad B_{\mathrm{v}} = B_{\mathrm{v}}\left(\boxed{\tau_{S_E}, \tau_{S_I}}\right)$$

Distributed forces:

$$\frac{\partial f_b}{\partial \tau} = -C\left(\vec{W} \cdot \vec{n}\right) \qquad f_v = \rho T \frac{W_m \boxed{\partial_m s}}{W} = f(\boxed{\boldsymbol{\omega}})$$

Loss coefficient



Correlations!

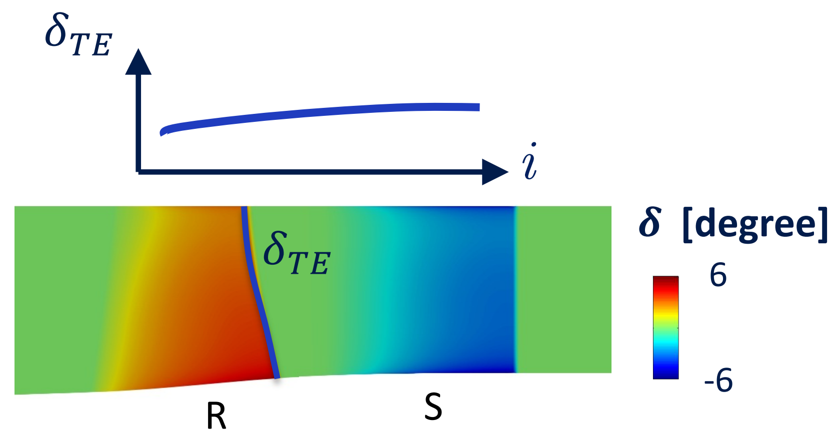
Deviation angle

Mean flow path = $f(cl, \boxed{\boldsymbol{\delta}})$

Correlations for δ and ω

Deviation angle δ → inviscid blade force

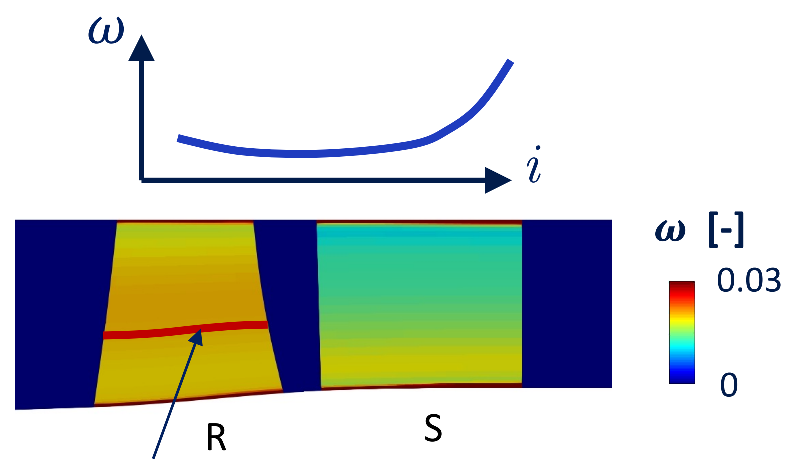
- δ_{TE} from cascade experiments (Lieblein)
- Linear variation with incidence around design conditions
- $\delta = \delta_{TE} \frac{\kappa_{LE} - \kappa}{\kappa_{LE} - \kappa_{TE}}$ ⟵ Blade angle



Loss coefficient ω → viscous blade force

- From cascade experiments (Lieblein)

⚠ Profile loss only



Outline

1

Viscous through-flow model

$$\frac{\partial U}{\partial t}+\frac{\partial (F-F_v)}{\partial x}+\frac{\partial (G-G_v)}{\partial r}=S$$

2

Model Assessment

Application to an axial LP compressor

3

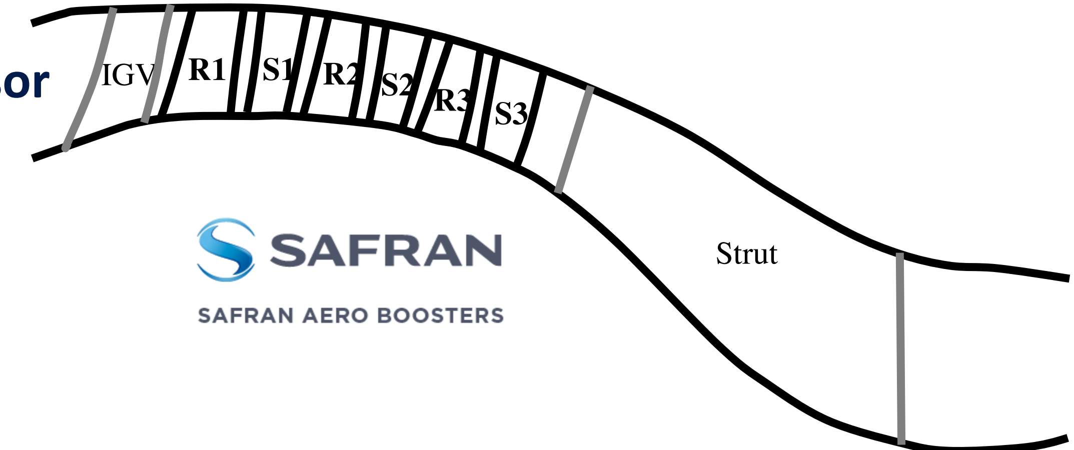
Geometrical variability

[SAB]

Closure model assessment

- Quantify closure **model errors**
- Simulations fed with **exact** distribution of $\boldsymbol{\delta_{TE}, \omega}$

Test-case : low-pressure compressor



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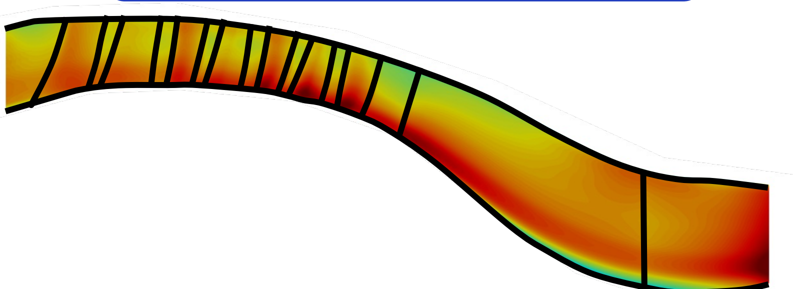
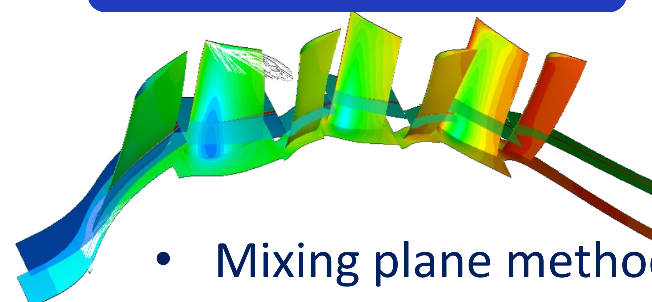
- Highly loaded
- High subsonic Mach number
- 3D modern blades

Through-flow (TF) simulations



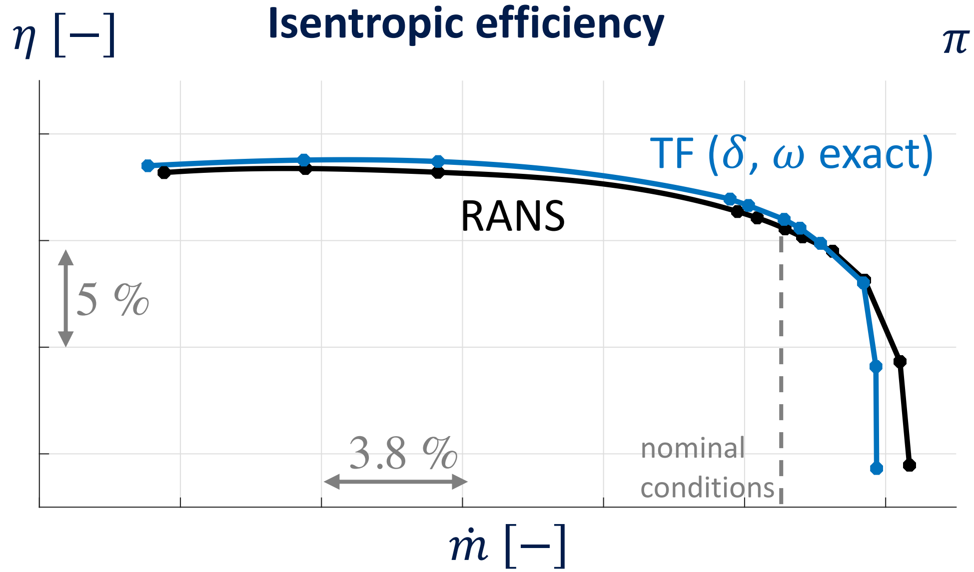
3D RANS simulations

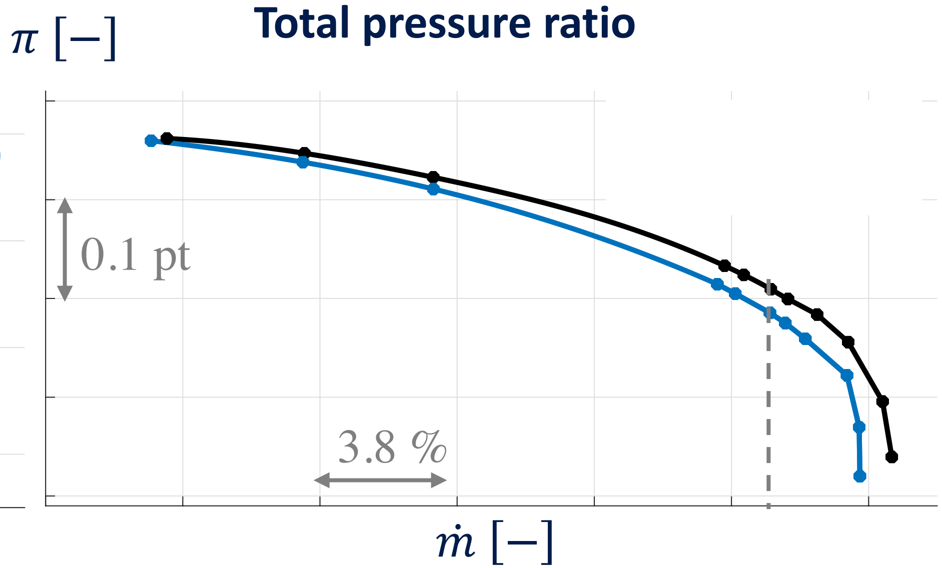


- Mixing plane method
- Steady, periodic flow

δ_{TE}, ω
distributions

Closure model assessment





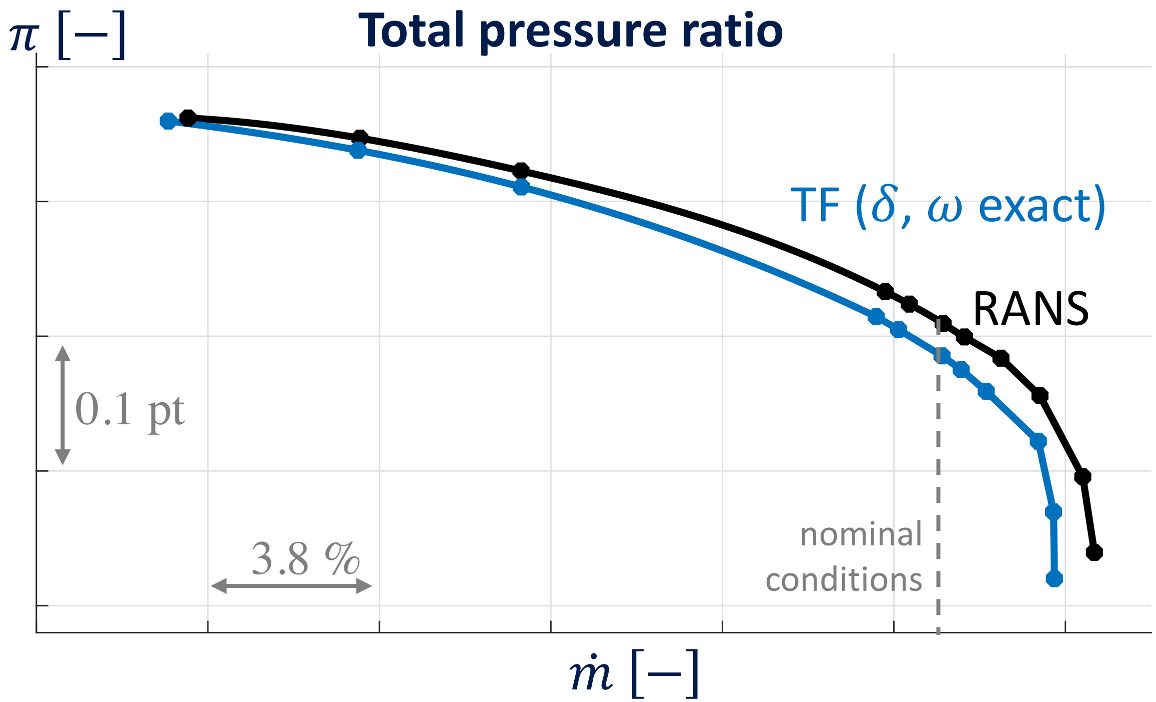
- Good prediction (low margin)
- 600 times faster
- Sources of erros: τ_{circ}, closure model form, δ_{TE} distribution, blockage assumption, turbulence model…

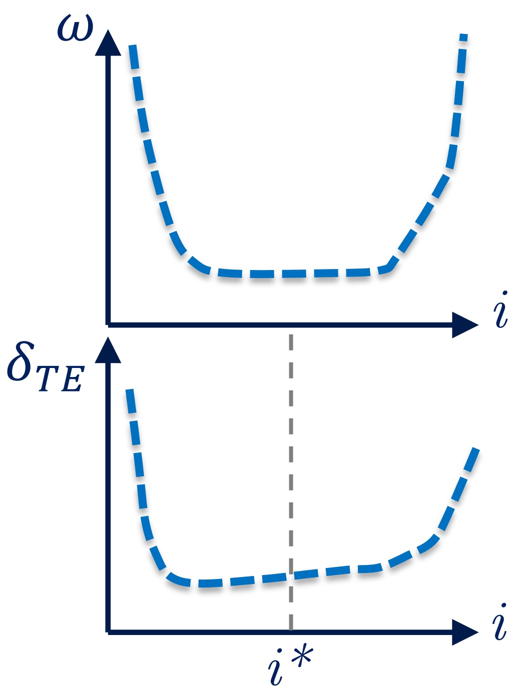
Model able to predict performance

But exact δ, ω unknown in practice…

Correlations assessment

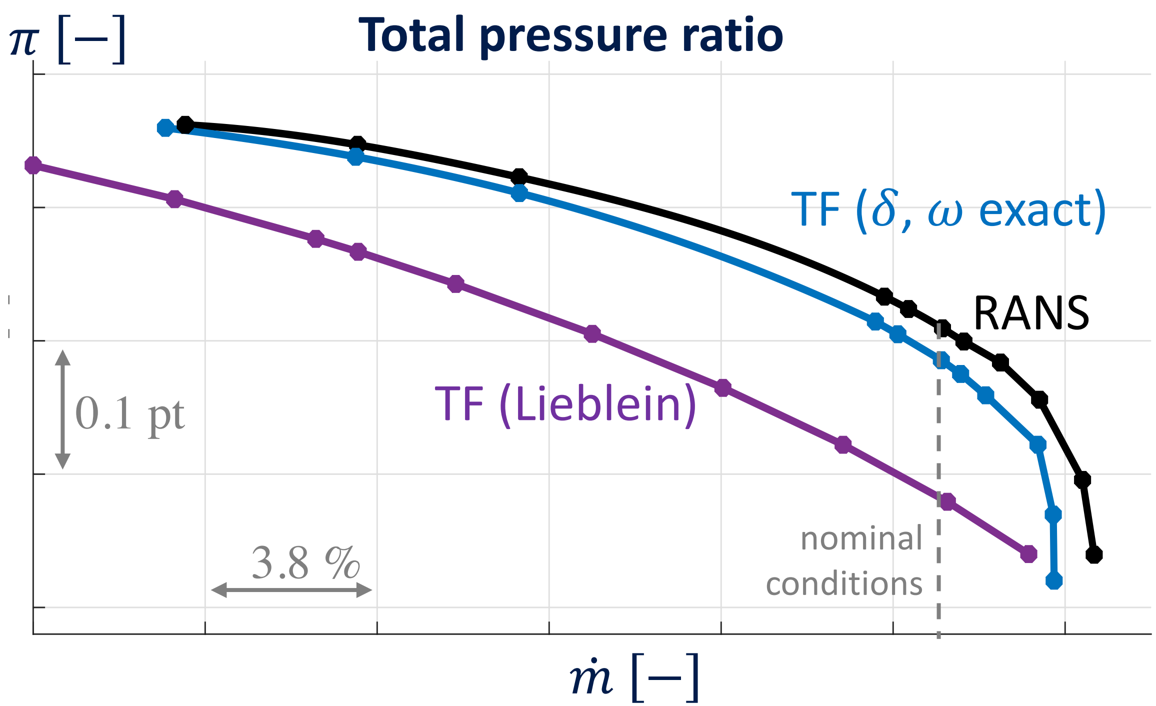
- Error quantification of **correlations** for $\boldsymbol{\delta}$, $\boldsymbol{\omega}$

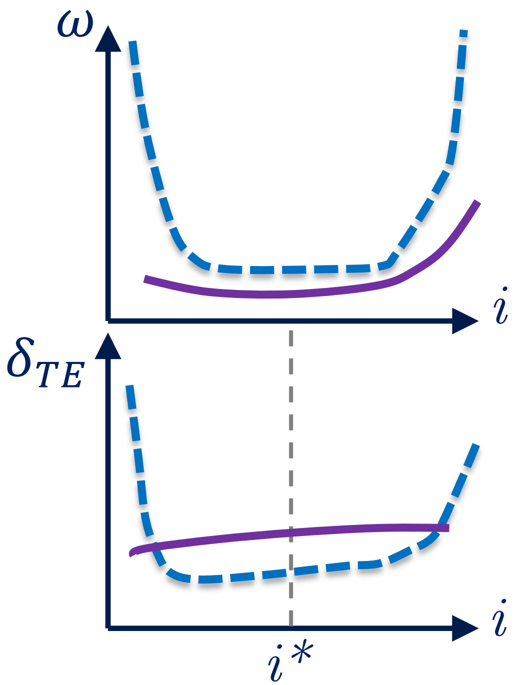




Correlations assessment

➤ Error quantification of **correlations** for $\boldsymbol{\delta}$, $\boldsymbol{\omega}$

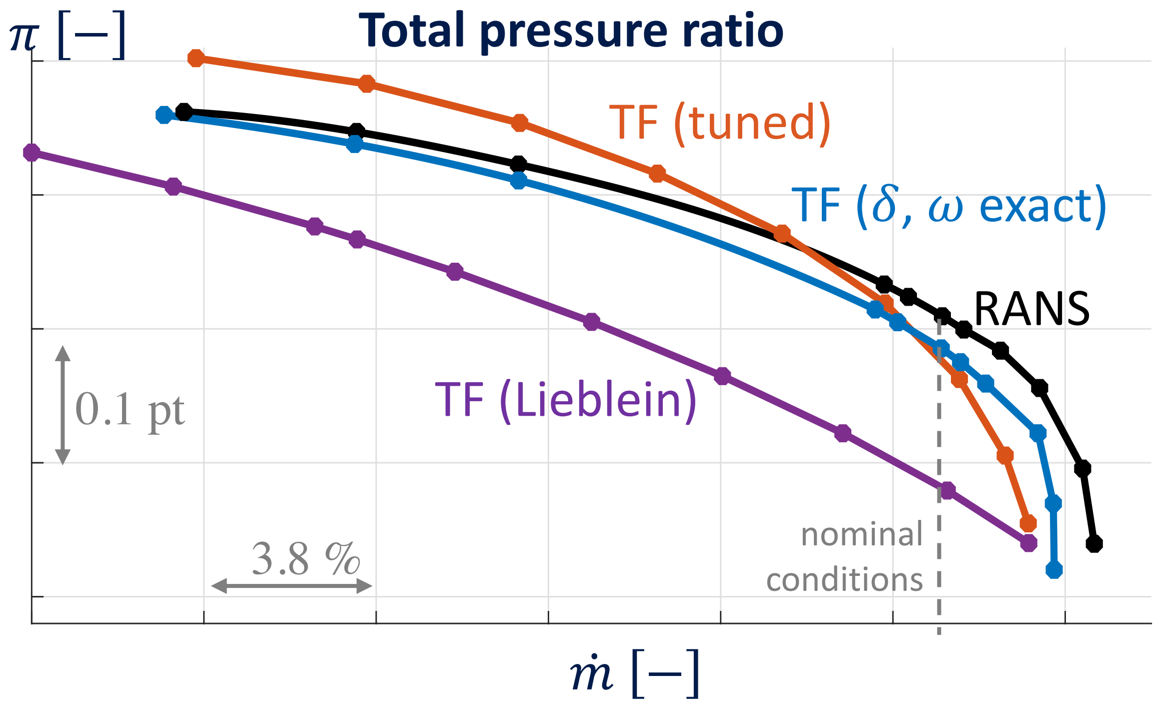


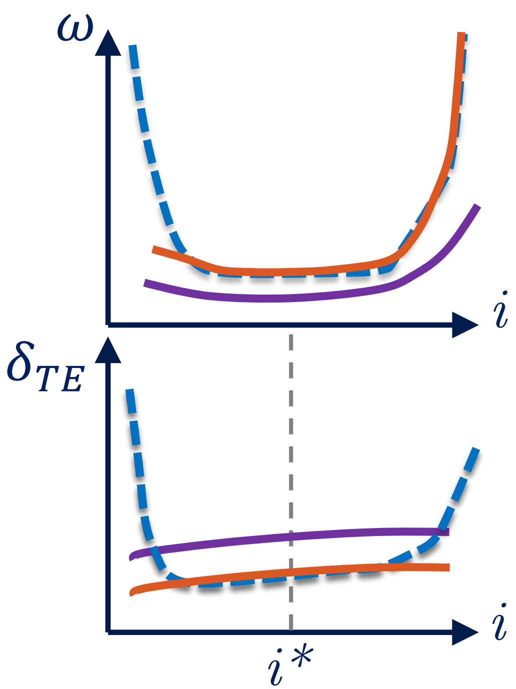


- Inaccurate when applied to the modern compressor

Correlations assessment

➢ Error quantification of **correlations** for $\boldsymbol{\delta}$, $\boldsymbol{\omega}$





- **Rotor** deviation angle **correction** → total pressure ratio improvement
- **Mach number effect** added to loss coefficient

Strong dependence of model prediction with respect to correlation accuracy

Outline

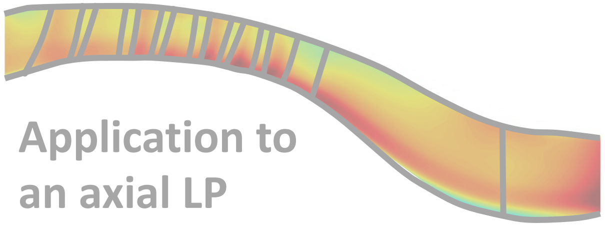
1

Viscous through-flow model

$$\frac{\partial U}{\partial t} + \frac{\partial (F - F_v)}{\partial x} + \frac{\partial (G - G_v)}{\partial r} = S$$

2

Model Assessment



Application to an axial LP compressor

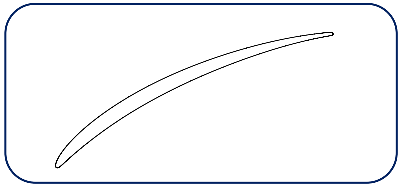
3

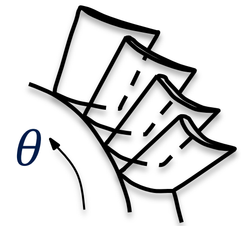
Geometrical variability

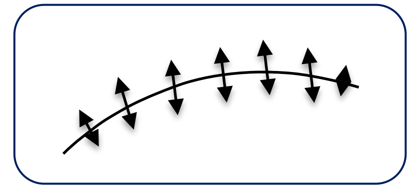
[SAB]

Geometry in through-flow model

$\mathrm{D}_t \boldsymbol{U}(r, \theta, x, t) = \boldsymbol{G}(U, r, \theta, x, t)$ $\xrightarrow{\theta\ \text{–averaging}}$ $\mathrm{D}_t \bar{\boldsymbol{U}}(r, \theta, x) = \bar{\boldsymbol{G}}(U, r, \theta, x)$



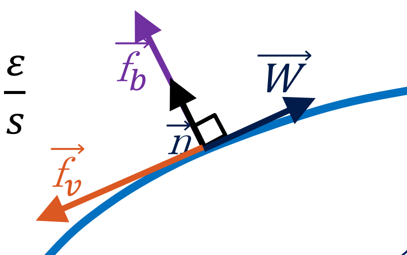




Camber line coordinates
&
Thickness distribution

Blade forces

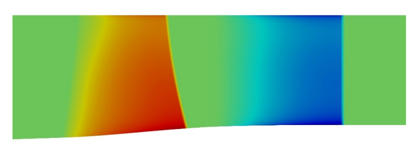
$$b = 1 - \frac{\varepsilon}{s}$$



Direct impact

Correlations

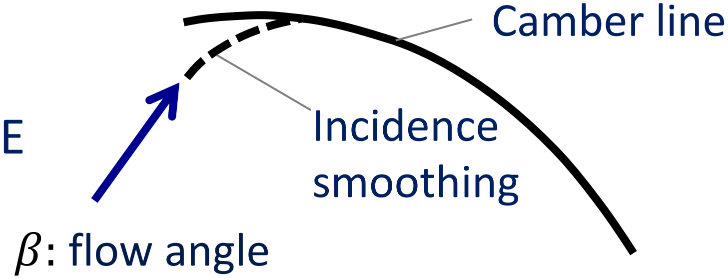
$\delta_{TE}, \omega, i* = f\left(\begin{array}{c}\text{geometry,}\\ \text{flow quantities}\end{array}\right)$

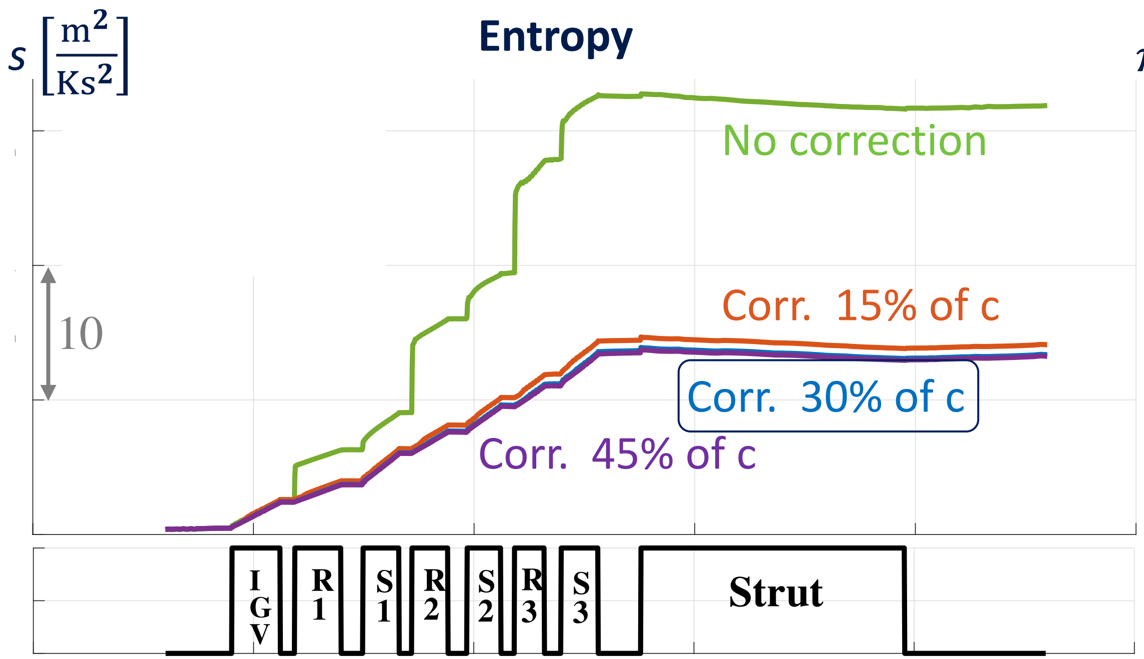


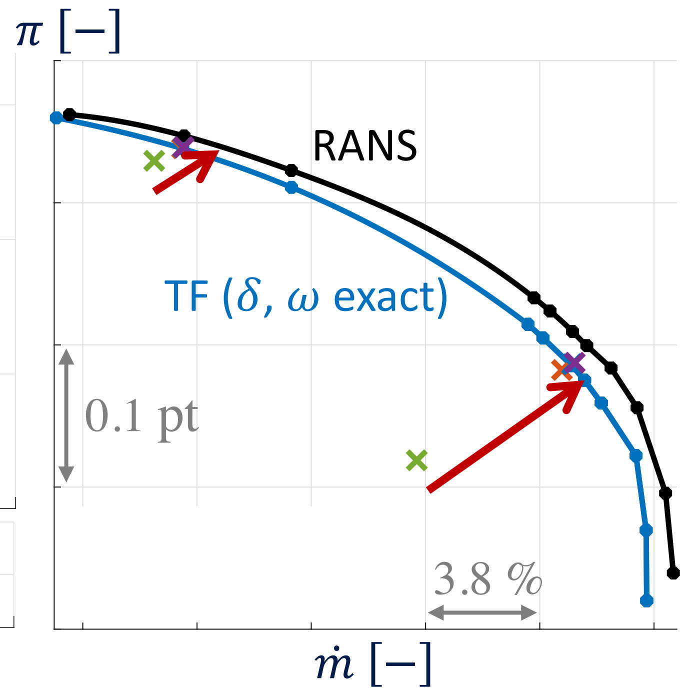
Indirect impact

Incidence correction

- Avoid flow angle discontinuity
- Modification of blade skeleton @ LE
- Unchanged correlation input

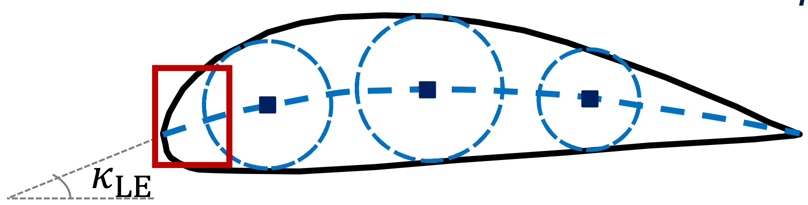


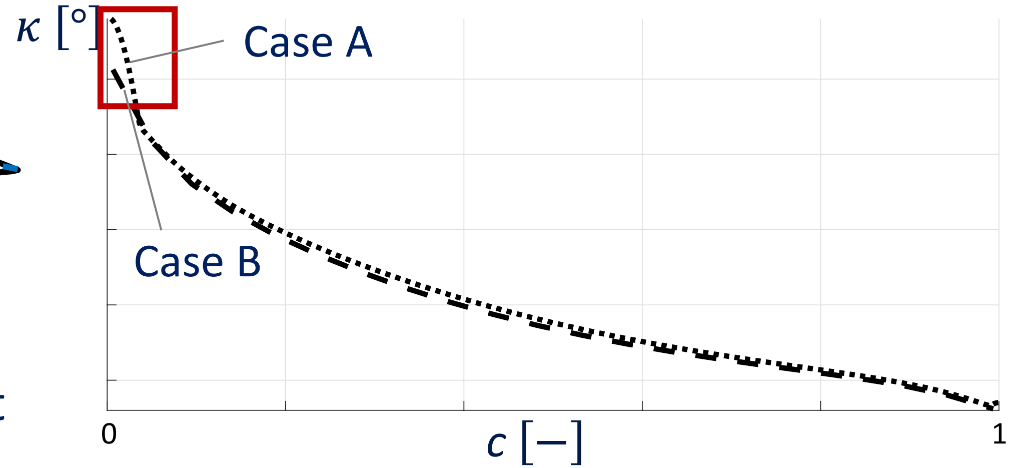




Incidence correction can smooth variability @ LE

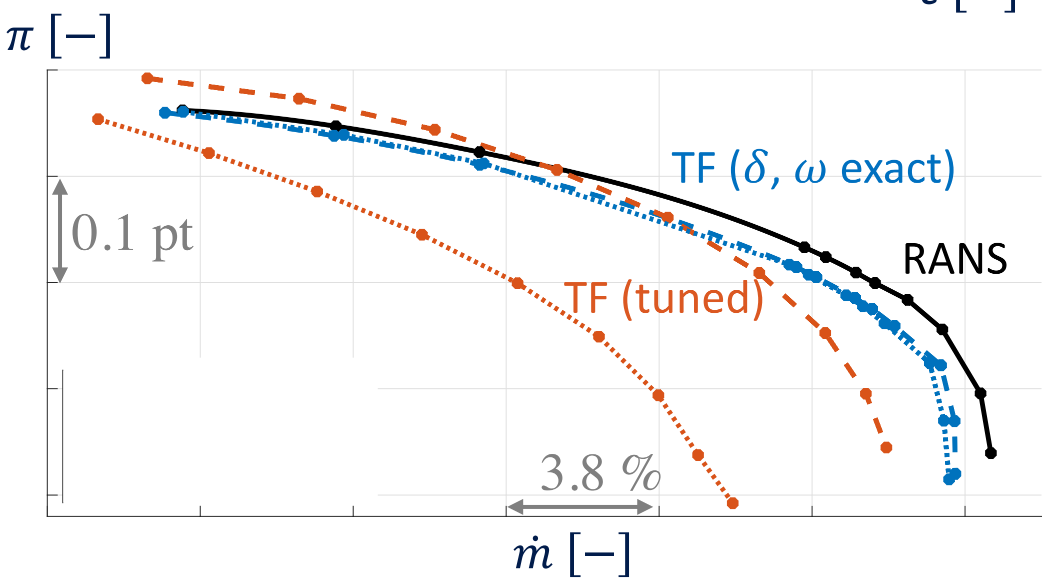
Camber line definition





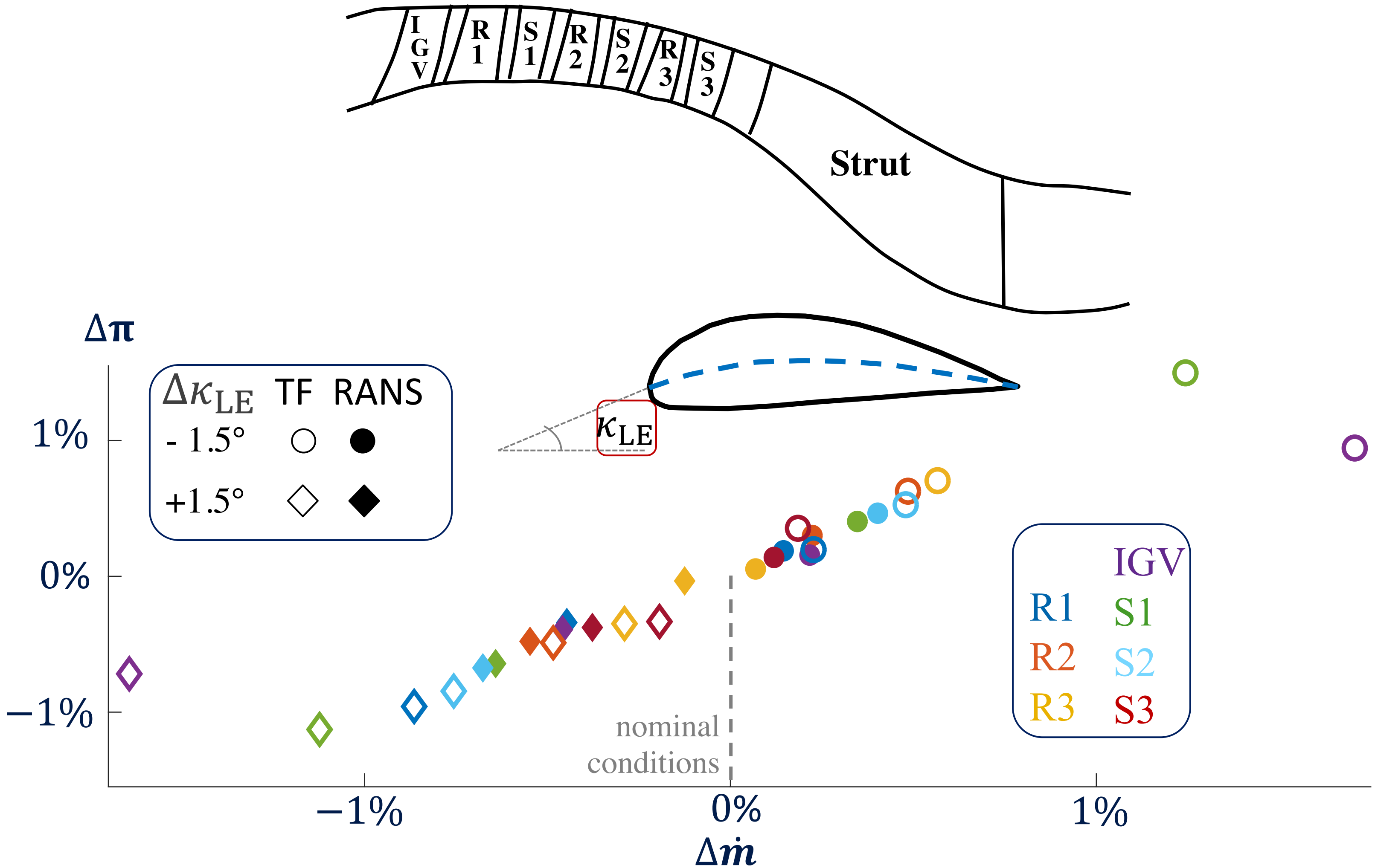
Definition not unique close to LE

→ Large impact on correlation input



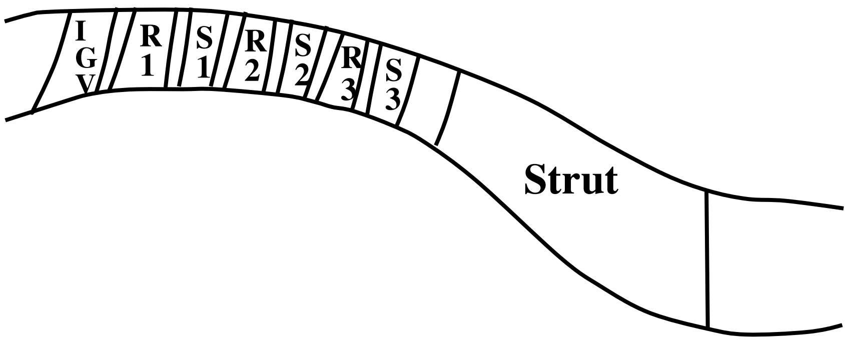
Strong dependence of model prediction with respect to LE blade angle

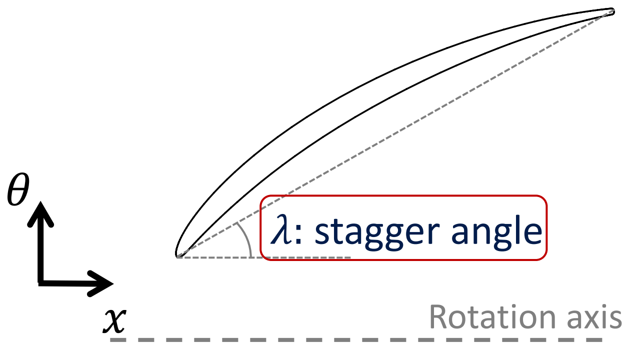
Applications: blade angle variability

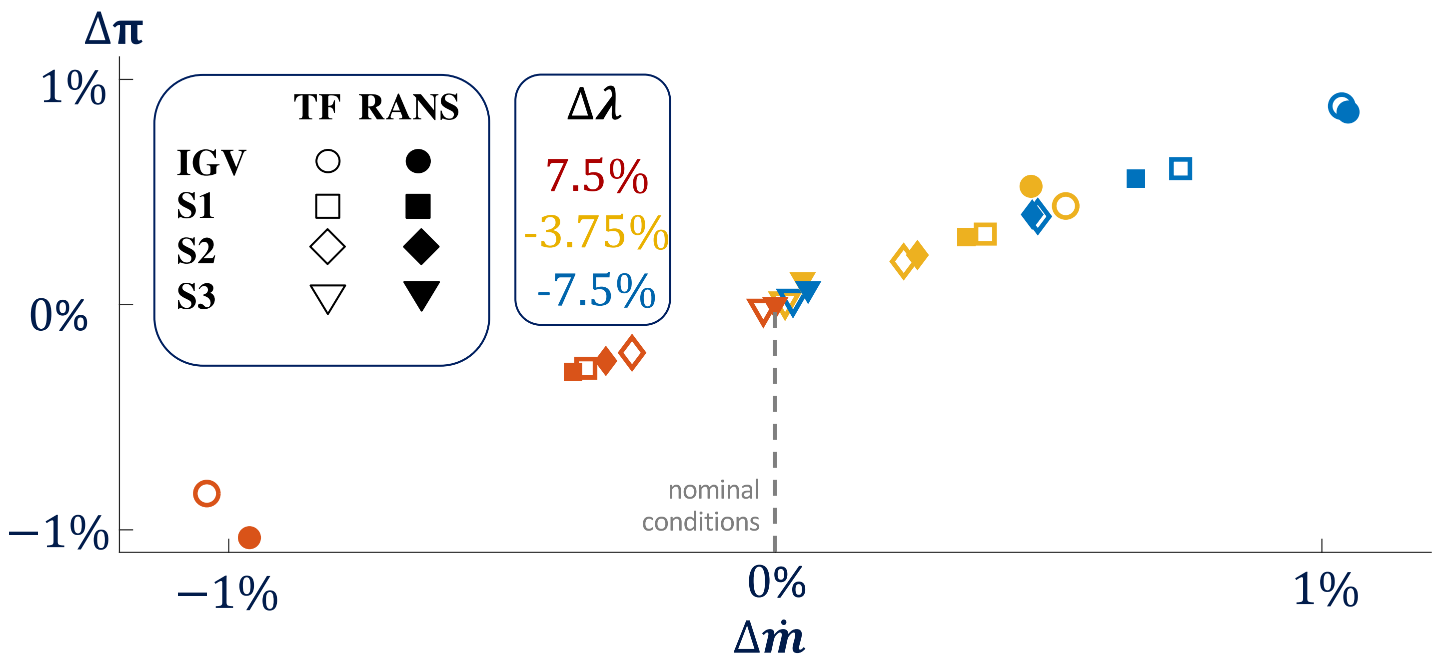


Overestimation of performance variability

Applications: stagger angle variability







Model able to predict performance variation

Conclusion

Through- flow model

- **Reliable low-fidelity** method
- **Good prediction** of performance
- Strong **dependence** between performance prediction and **correlation accuracy**
- Promising approach to drastically **reduce CPU cost** compared to 3D RANS for multi-fidelity approach and UQ

Geometrical variability

- Modeling aspects (incidence correction and camber line definition) **smear variability propagation @ LE**
- Global good agreement for other **performance variation**

Future work

- Correlation **improvement** @ high incidence
- Thorough analysis of **geometric variability propagation**
- **Strength and weakness** of the model

Acknowledgement

Funding for this research is provided by the Walloon region, under grant no. 7900 in the frame of the project MARIETTA

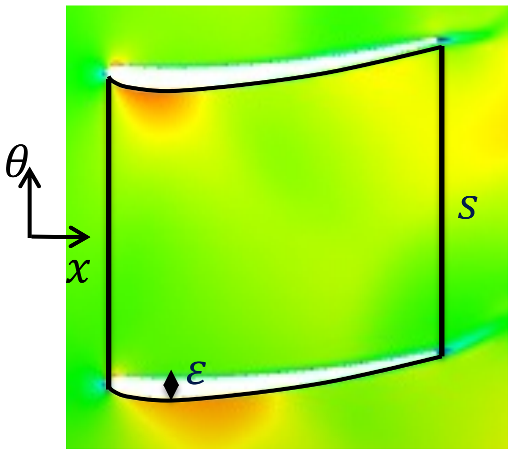
BACK-UP

Viscous through-flow model

Circumferential averaged Navier-Stokes equations:

Conservative variables

$$\frac{\partial \boldsymbol{U}}{\partial t} + \frac{1}{b}\frac{\partial b\overbrace{(\boldsymbol{F} - \boldsymbol{F}_v)}^{x\text{-fluxes}}}{\partial x} + \frac{1}{b}\frac{\partial b\overbrace{(\boldsymbol{G} - \boldsymbol{G}_v)}^{r\text{-fluxes}}}{\partial r} = \boxed{\boldsymbol{S}}$$



Blockage factor

$$b = 1 - \frac{\varepsilon}{s}$$

- Reynolds stress
- Inviscid blade force
- Viscous blade force
- Axisymmetric source terms

Non-intrusive formulation for CFD solver:

$$\frac{\partial \boldsymbol{U}}{\partial t} + \frac{\partial(\boldsymbol{F}-\boldsymbol{F}_v)}{\partial x} + \frac{\partial(\boldsymbol{G}-\boldsymbol{G}_v)}{\partial r} = \boldsymbol{S} + \boxed{\frac{(\boldsymbol{F}_v-\boldsymbol{F})}{b}\frac{\partial b}{\partial x} + \frac{(\boldsymbol{G}_v-\boldsymbol{G})}{b}\frac{\partial b}{\partial r}}$$

Blockage factor terms (known)

Viscous through-flow model: ASTEC

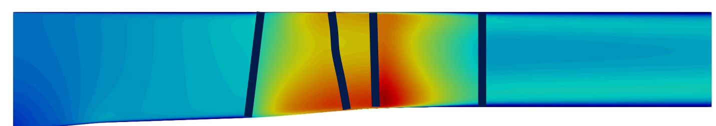
Methodology:

- Time-marching
- Non-intrusive

Initialization

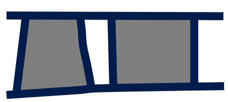
elsA

- Compressible, viscous flow
- Conservative variables: $\rho, \rho\boldsymbol{V}, \rho E$
- Turbulence model
- Axisymmetric source terms



ASTEC

- Python interface
- Computation:
  - Blade forces
  - Blockage factor terms



Iteration t
Convergence?

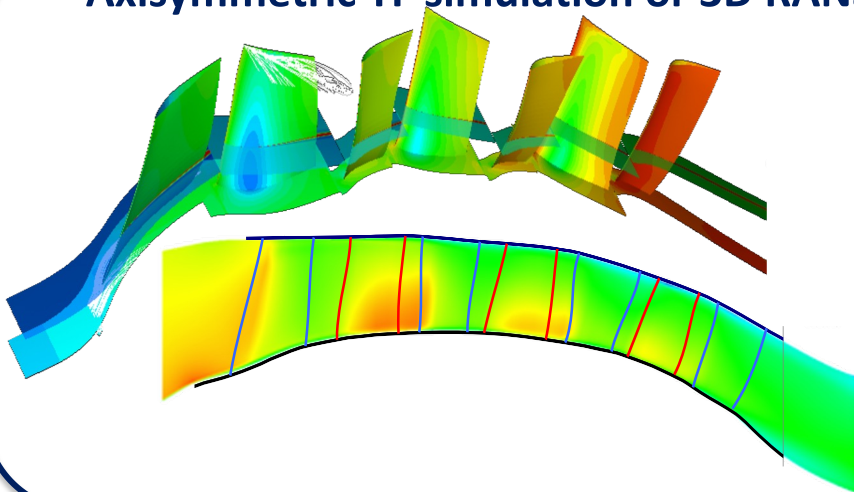
No

yes

Outputs

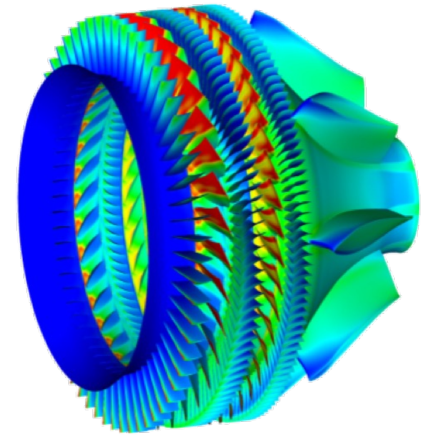
Relevance

Axisymmetric TF simulation or 3D RANS

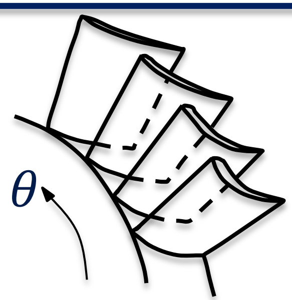


≠

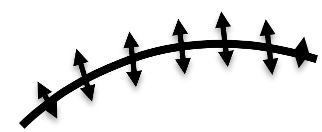
Mistuning



But



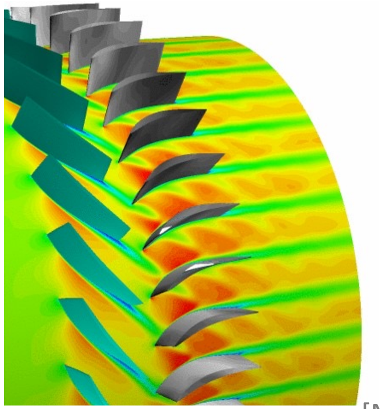
Averaged blade row



- Recurring patterns (tolerances, blade production, …)
- Low deformation distribution

Correlations assessment

➢ Error quantification of **correlations** for $\boldsymbol{\delta}$, $\boldsymbol{\omega}$

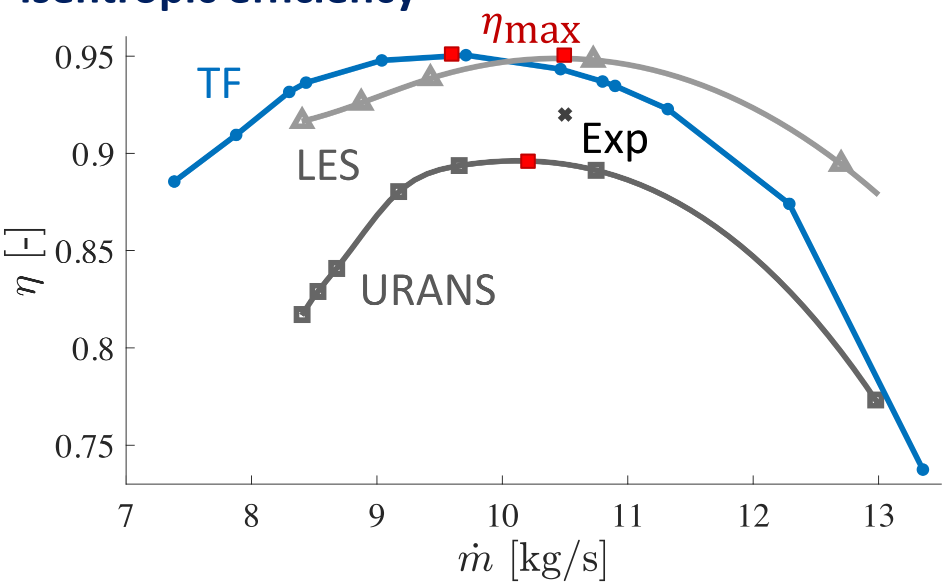


[Moreau 2019]

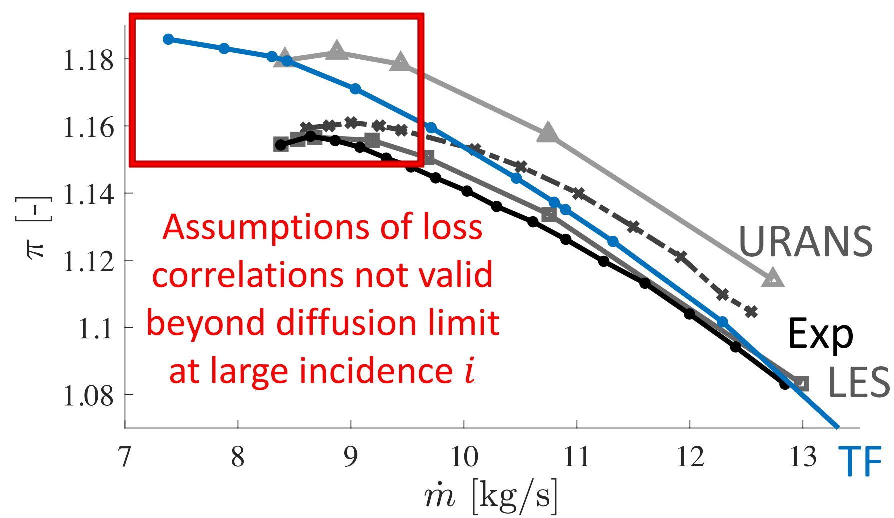
Test-case 2: CME2

- Research compressor designed by Safran Aircraft Engines
- Low speed flow
- Correlations calibrated at these conditions

Isentropic efficiency



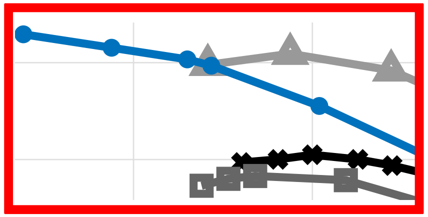
Total pressure ratio



Source: [Gourdain 2015]

CME2: Diffusion limit

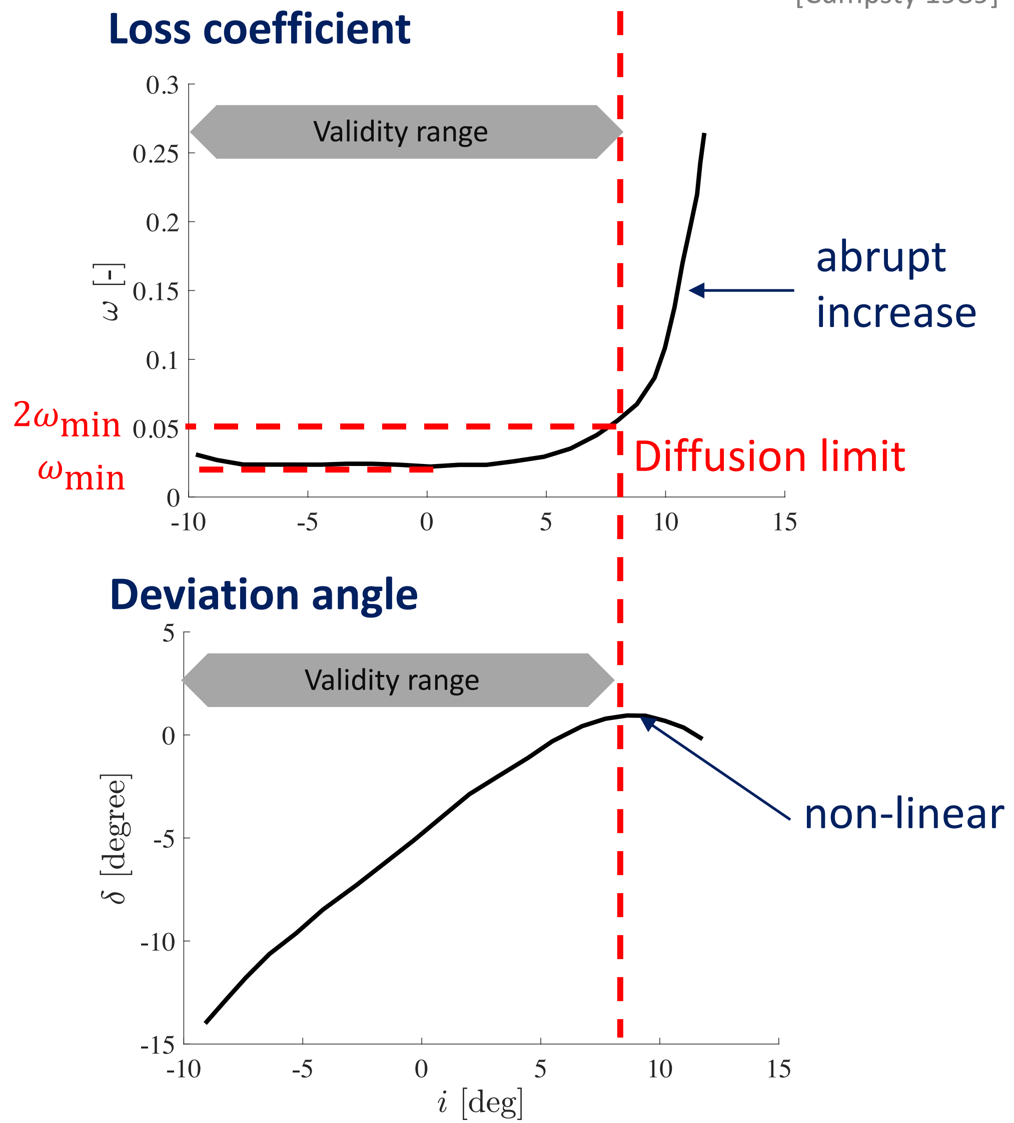
Total pressure ratio



Assumptions of loss correlations not valid beyond diffusion limit at large incidence i

Measurements of C4-series cascade (M = 0.4)

[Cumpsty 1989]



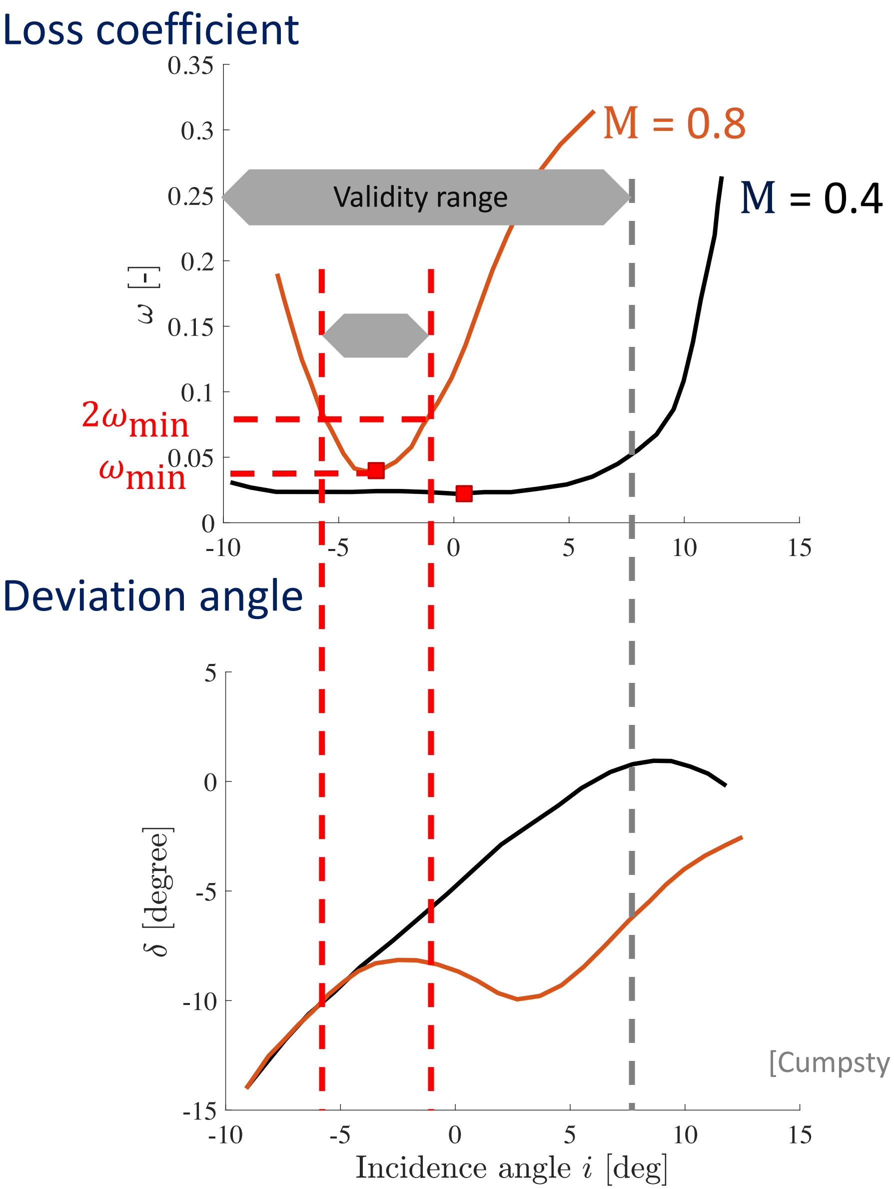
Modern compressor: comparison to RANS

Measurements of C4-series cascade

Impact of Mach number

- Minimum-loss incidence angle shifted
- Increase of $\omega_{\min}$
- Narrow range of validity
- Inconsistency between loss validity range and deviation linear range

Correlations **not calibrated** for these **flow conditions**



[Cumpsty 1989]