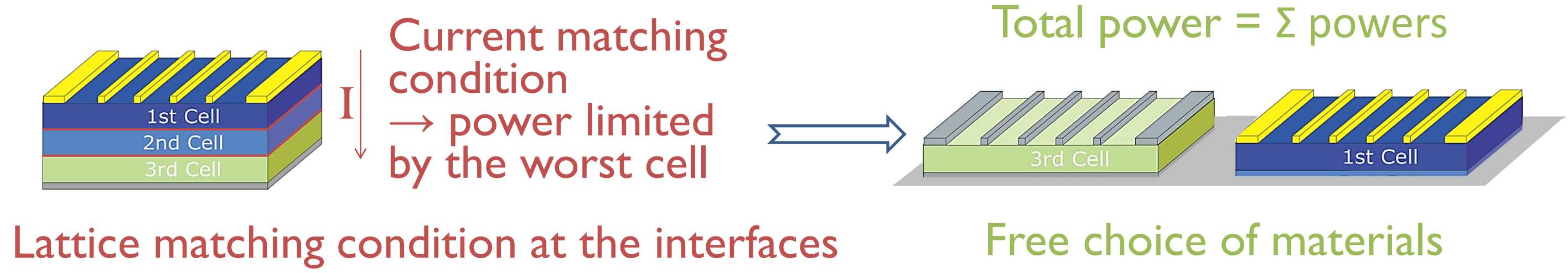


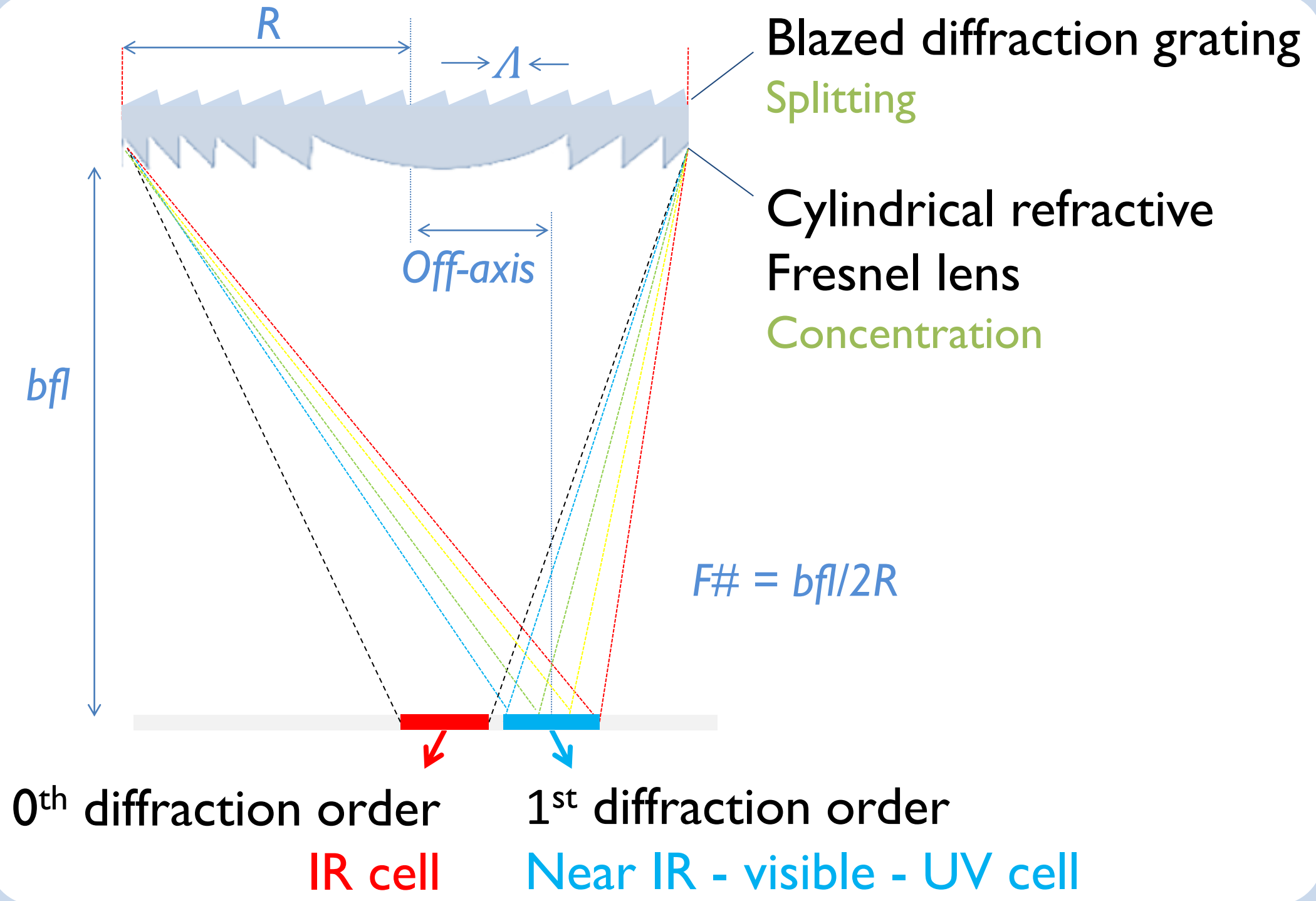
Context : satellites needs

Solar concentration : # solar cells/m² ↓ → cost ↓
Space environment → specific thermal conditions → desired solar concentration about 10x

Spectrum splitting

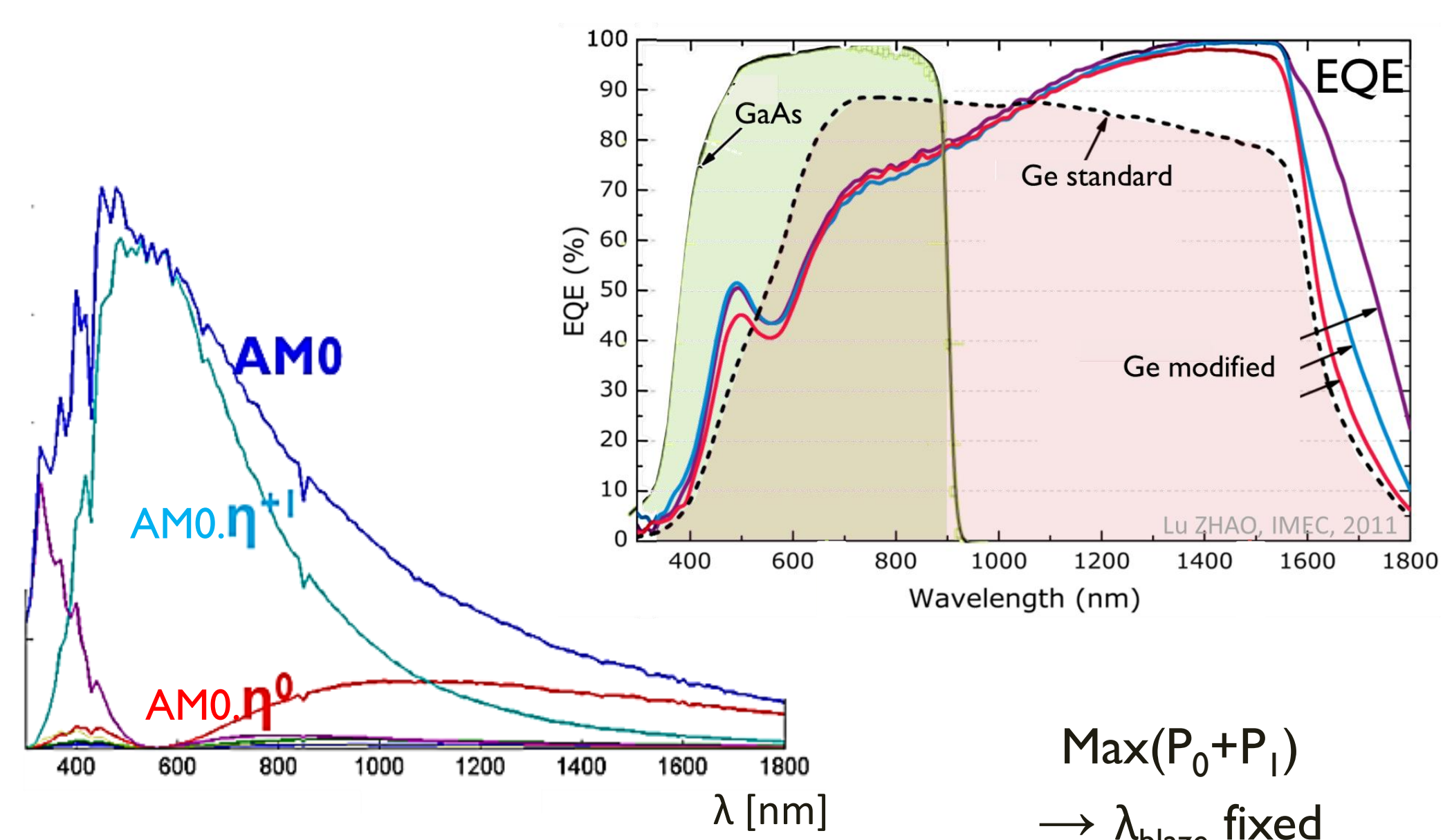


Combination



Design and optimization : a lot of constraints → a compromise must be found

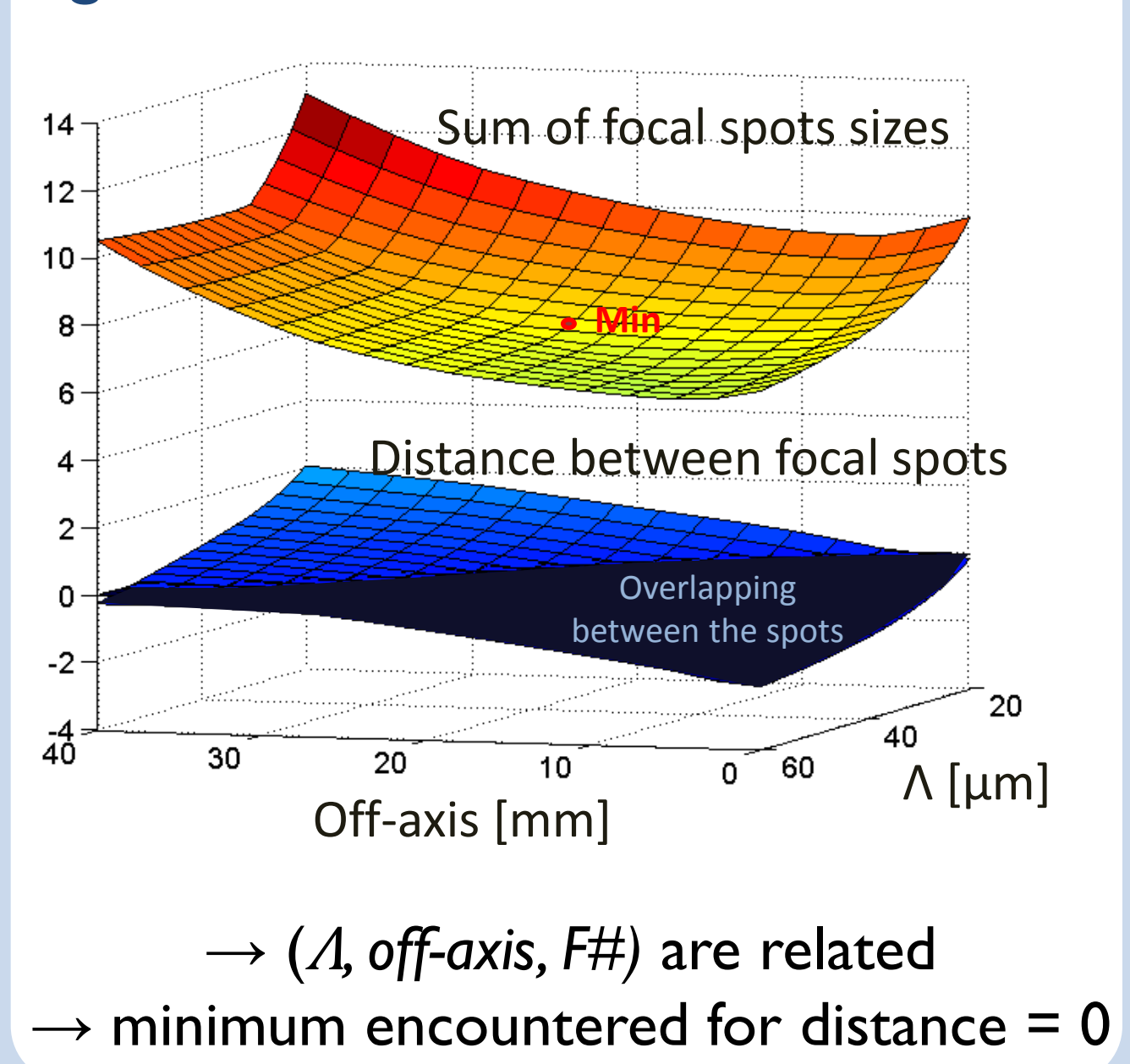
Maximum total output power



$$P_0 + P_1 = \int AM0. \eta_{diff}^0(\lambda_{blaze}) \cdot \frac{\lambda \cdot q}{h \cdot c} \cdot EQE_{Ge} d\lambda + \int AM0. \eta_{diff}^1(\lambda_{blaze}) \cdot \frac{\lambda \cdot q}{h \cdot c} \cdot EQE_{GaAs} d\lambda$$

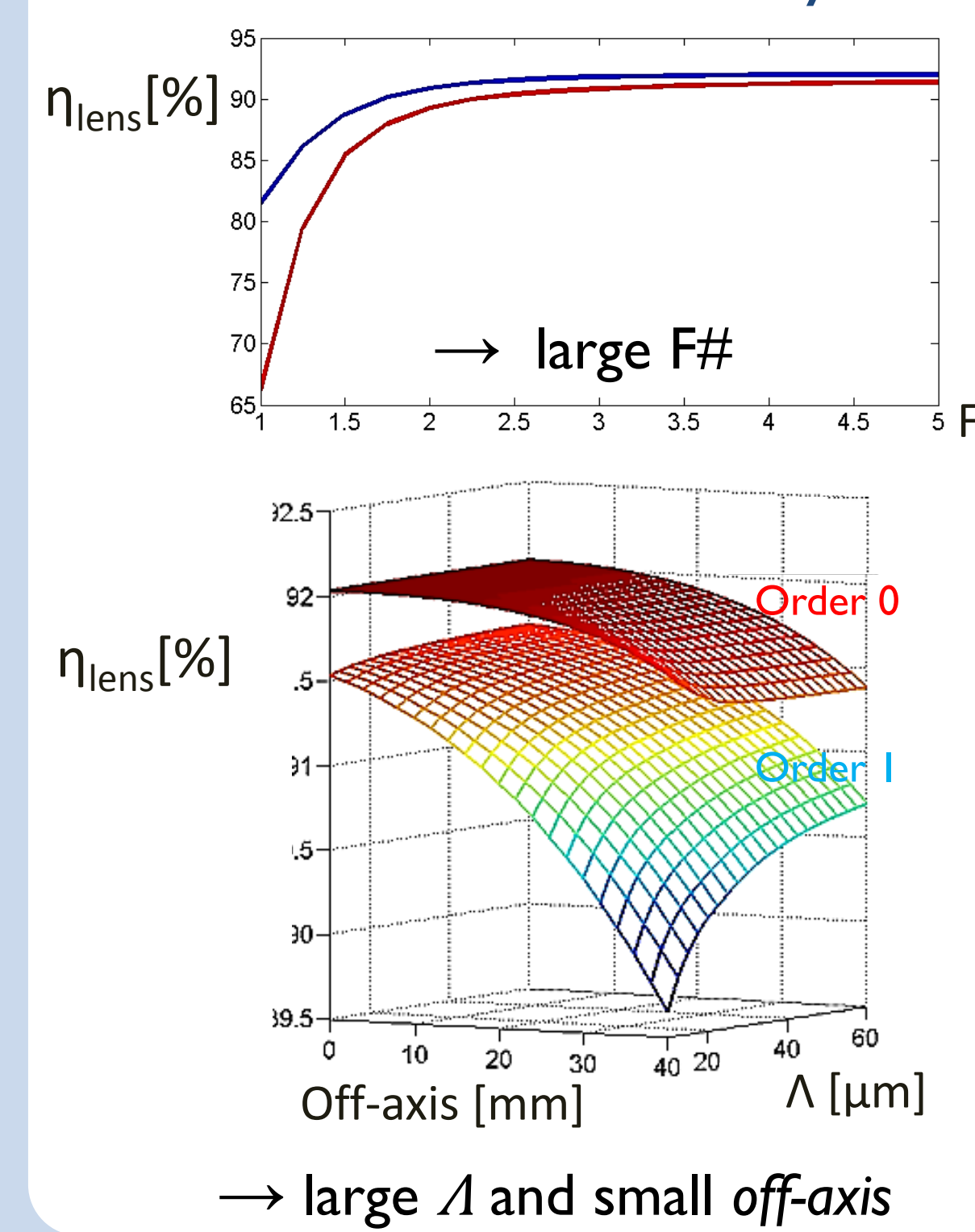
Max 2nd order diffracted light collection recovery → large Δ

Maximum geometrical concentration ratio

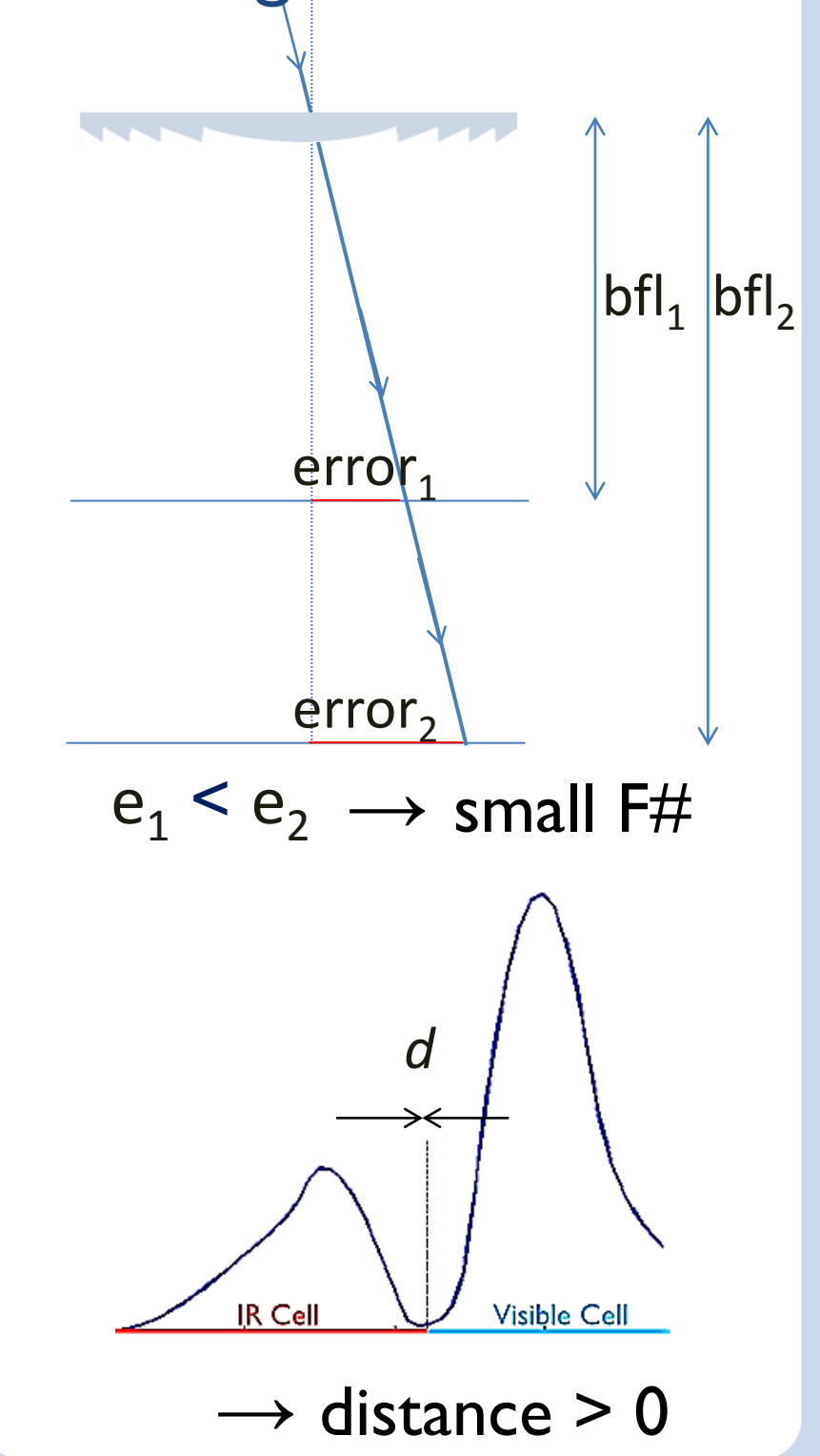


Thermal constraints → C_{geo} is limited

Maximum lens transmission efficiency

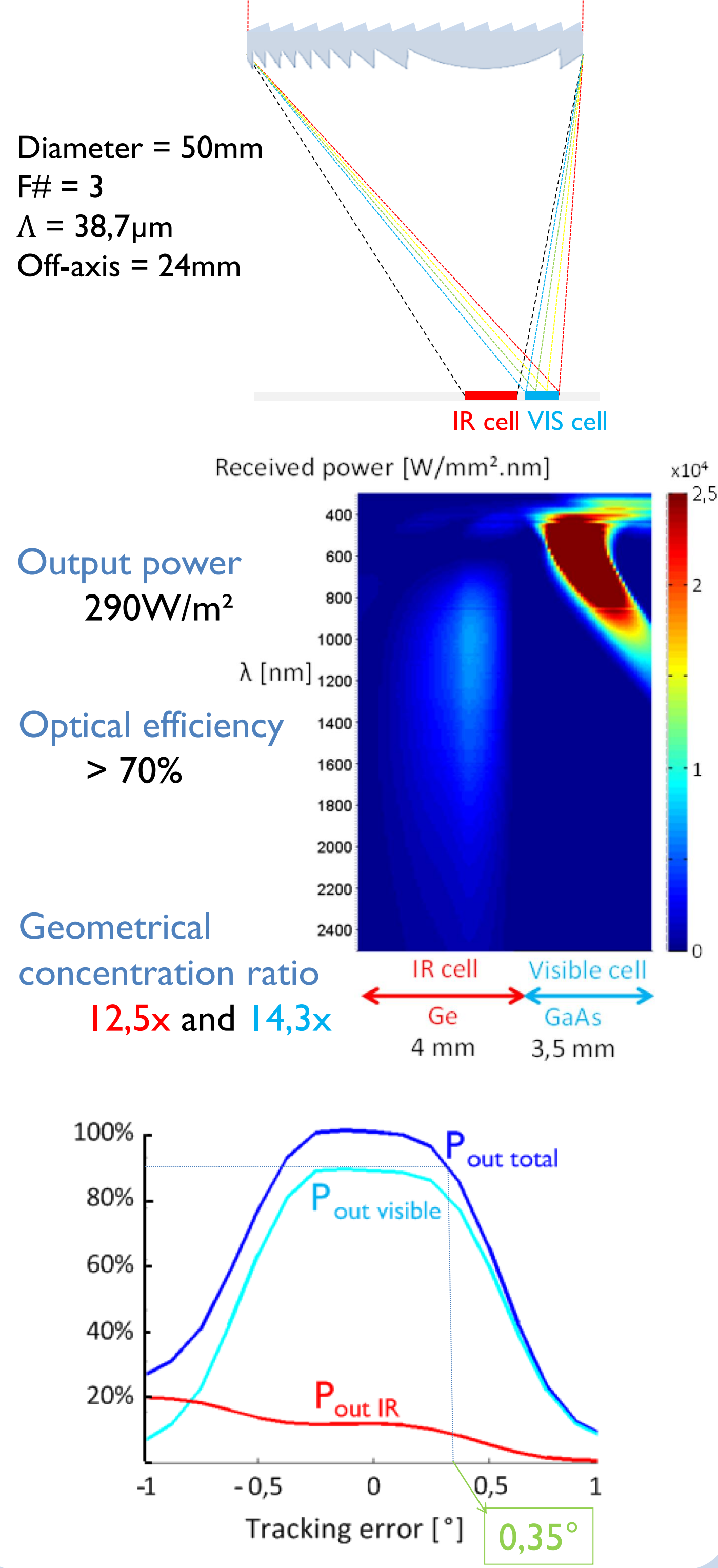


The best tracking tolerance

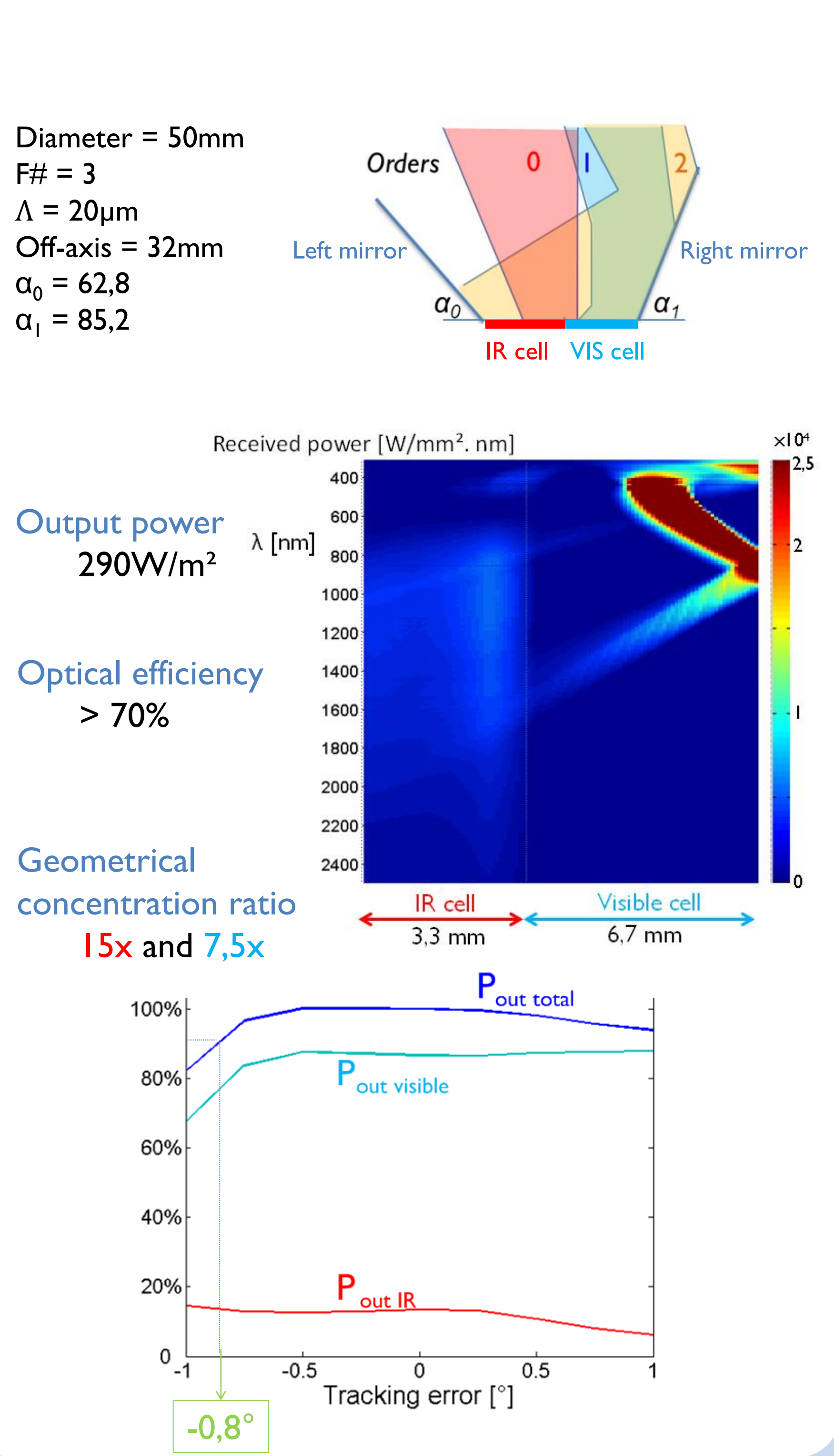


Different configurations have been studied : the results are promising

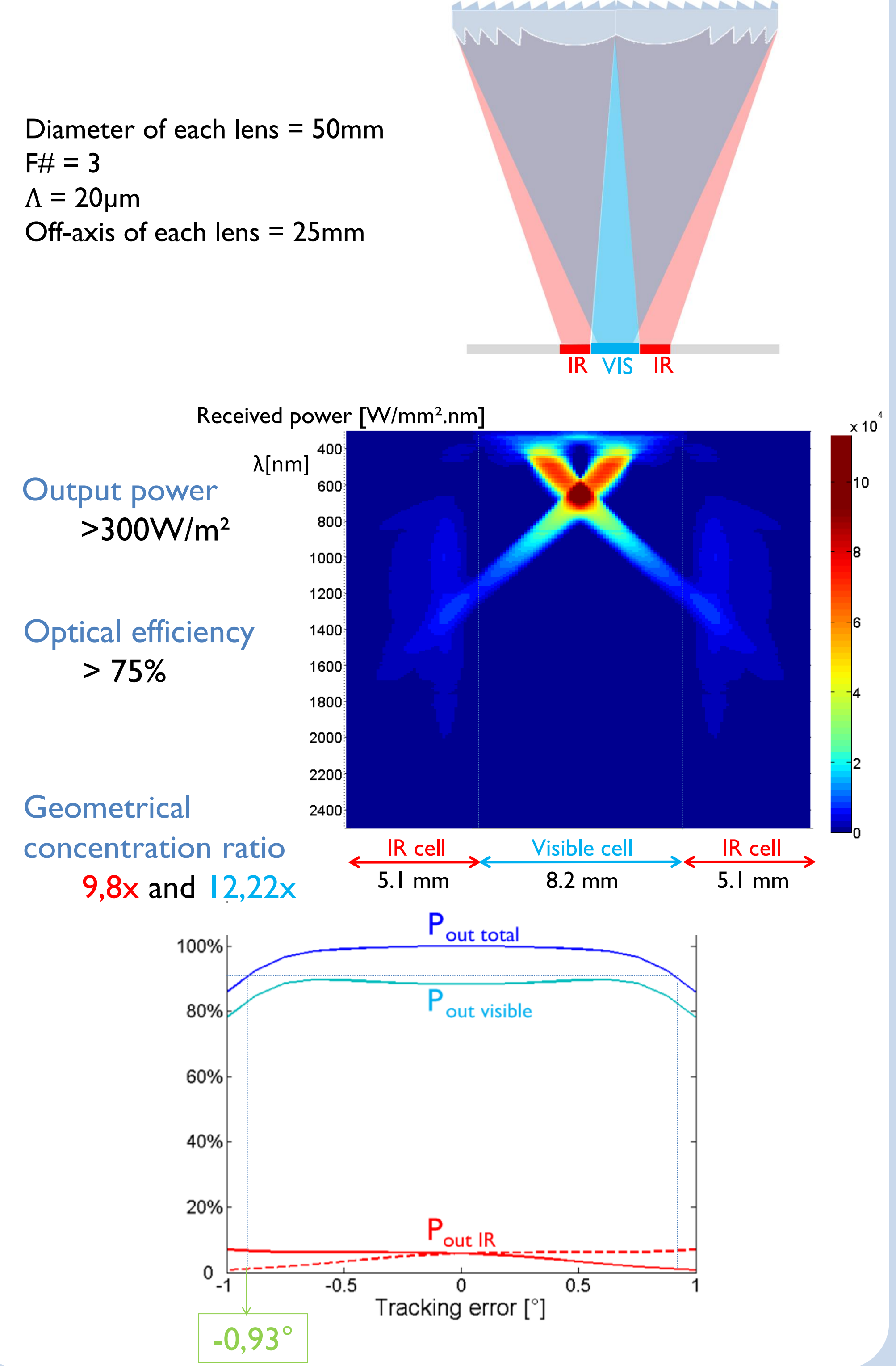
First configuration



With secondary concentrators



Symmetrical design



Next steps : thermal simulation and validation

- Cell efficiency ↓ when the temperature ↑
- Space: no convection → heat transfer is difficult
→ hot spots appear on solar cells

⇒ A thermal simulation is necessary to know the maximal concentration that can be achieved without damaging the cells

Hypothesis : Sun divergence (±0,26°), AM0 spectrum, cell temperature = 65°C. Fresnel reflections and shadowing of the grooves are taken into account, diffraction efficiencies of the grating are computed from the scalar diffraction theory. The shapes of the Fresnel lens (in DC93-500 silicone) and of the diffraction grating are ideal (no draft angles, no rugosity, ...). Light scattering is not included in the model.